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REPORT NO. RD-TR-65-6

# TABULATED STABILITY DATA FOR A SERIES OF RING TAIL BODY CONFIGURATIONS AT MACH NUMBERS FROM 0.8 TO 4.5

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by

Donald V. Rubin

April 1965



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# TABULATED STABILITY DATA FOR A SERIES OF RING TAIL BODY CONFIGURATIONS AT MACH NUMBERS FROM 0.8 TO 4.5

by
Donald V. Rubin

DA Project No. 1S222901A206

AMC Management Structure Code No. 5221.11.148

Aerodynamics Branch
Advanced Systems Laboratory
Directorate of Research and Development
U. S. Army Missile Command
Redstone Arsenal, Alabama

#### **ABSTRACT**

The static stability characteristics of ring tails on a body of revolution has been investigated during a series of wind tunnel tests. Mach numbers varied from 0.8 to 4.5, and ring diameters varied from 1.25 to 2.50 calibers. This report presents a tabulation of the basic results obtained from the tests.

#### SYMBOLS AND MODEL NOMENCLATURE

CAB	Base Axial-Force Coefficient, Base Axial Force
CAF	Forebody Axial Force Coefficient, Forebody Axial Force
CM	Pitching Moment Coefficient, Pitch Moment About Nose  qS, Db
CN	Normal Force Coefficient, Normal Force
$D_{b}$	Reference diameter-body maximum diameter
M	Mach number
Q	Dynamic pressure, psi
RN	Reynolds number per in. X 10 <sup>-6</sup>
S	Reference area, $\frac{D_b^2}{4}$
ALP	Angle of attack
В	Body
T	Ring Tail Configuration
x	Ring Tail Position, where:
	<ul> <li>X1 - Ring trailing edge 0.5 caliber forward of model base</li> <li>X2 - Ring trailing edge at the model base</li> <li>X3 - Ring trailing edge 0.5 caliber aft of model base</li> <li>X4 - Ring trailing edge 0.9 caliber aft of model base</li> <li>X5 - Ring trailing edge 0.25 caliber aft of model base.</li> </ul>

#### 1. Introduction

The Aerodynamic Branch of the Advanced Systems Laboratory is currently engaged in a supporting research program that primarily involves base drag reduction.\* One promising method of base drag reduction is by the favorable interference induced by a ring tail. However, before investigating the effect of ring tails on base drag, it should first be determined if any undesirable stability characteristics are present.

A preliminary wind tunnel investigation was made to explore the force and moment characteristics of a systematic series of ring tail-body combinations. The purpose of this report is to present the experimental results obtained at Mach numbers from 0.8 to 4.5 and through an angle-of-attack range that varied from -4 degrees to 6 degrees.

Analysis of these results is in preparation. 1, 2

#### 2. Apparatus and Procedure

The transonic portion of the tests was conducted in the Aero-dynamic Wind Tunnel, Transonic (T1) of the Propulsion Wind Tunnel Facility, Arnold Engineering Development Center (AEDC), Arnold Air Force Station. The supersonic portion of these tests was conducted in Supersonic Wind Tunnel No. 1, Ballistic Research Laboratories (BRL), Aberdeen Proving Ground, Maryland.

The Transonic (T1) facility is an open-circuit, continuous-flow wind tunnel capable of operation at Mach numbers from 0.5 to 1.5. The test section is 12 inches square by 37.5 inches long and has four parallel perforated walls.<sup>3</sup>

<sup>\*</sup>This program is being conducted as a part of SR-Project, Base Drag Reduction, Code 5210.11.148.

<sup>&</sup>lt;sup>1</sup> Paul R. Connolly, "The Zero-Lift Foredrag and Body Base Drag Coefficients of a Series of Ring Tail-Struct-Body Configurations at Mach Numbers from 0.8 to 4.5." Army Missile Command RF-TR-65-8.

<sup>&</sup>lt;sup>2</sup> Charles E. Brazzell, "Longitudinal Stability Characteristics of a Series of Ring Tails on Bodies of Revolution at Mach Numbers From 0.8 to 4.5." Army Missile Command RF-TR-65-7.

<sup>&</sup>lt;sup>3</sup> Arnold Engineering Development Center, Propulsion Wind Tunnel Facility, Vol. 3, July 1963, Test Facilities Handbook.

The Supersonic Wind Tunnel No. 1 is a closed-circuit, continuousflow, variable-density facility with a flexible nozzle for obtaining a Mach number range of from 1.5 to 5.0. The test section is 13 inches wide by 15 inches high. 4

The same model was tested at AEDC and at BRL. The model was constructed so that components could be interchanged to test various body and ring tail combinations. The test bodies were 1.15 inches in diameter with a 4-caliber ogive nose and a total length of 10 calibers (Figure 1). Ring tails are attached to the bodies with either the four round support posts (Figure 2) or with the four faired support fins (Figure 3).

Five wind tunnel test programs were conducted. Table 1 lists these programs and shows when and where each body tail configuration was tested. (Body Bl data from these five separate tests are shown in Table 3.) The body and total configuration data may be matched by referring to Table 1.

The model was sting mounted in the test section on a sixcomponent internal strain gage balance. Base pressure measurements were obtained from a pressure transducer connected to two pressure orifices located in the plane of the body base. The angle of attack range was generally from -4 degrees to 6 degrees, and the Mach number range for the combined tests extended from 0.8 to 4.5.

The calculated inaccuracies in the test data which take into account accuracy in the balance measurements and the tunnel test conditions are as follows:

AEDC	Transon	ic	(T)	١
------	---------	----	-----	---

Moo	CN	СМ	CAB	CAF
0.8	±0.01	±0.026	±0,008	±0.012
1.0	±0.008	±0.022	±0.006	±0.010
1.5	±0,007	±0.017	±0,005	±0,008

<sup>&</sup>lt;sup>4</sup> J. C. McMullen, Ballistic Research Laboratories, Wind Tunnel Testing Facilities at the Ballistic Research Laboratories, July 1960.

BRL Supersonic Wind Tunnel No. 1

M <sub>oo</sub>	CN	CM	CAB	CAF
1.75	±0.008	±0.024	±0.005	±0.005
3.0	±0.008	±0.021	±0.004	±0.004
4.5	±0.009	±0.017	±0.005	±0.005

Configuration T13R has the same dimensions as T13 except the ring has been reversed to give internal compression instead of expansion.

#### 3. Results

An index of the tabulated data is shown in Table 2. The aerodynamic coefficients for the various configurations are presented in tabulated form in Tables 3 through 54.

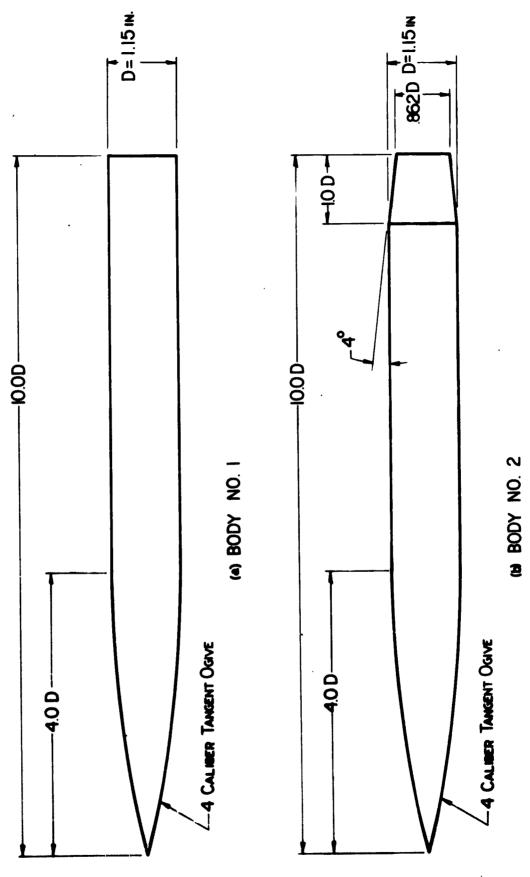
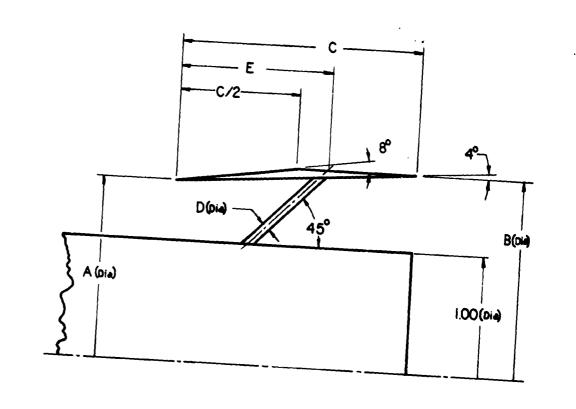


Figure 1. Body Configuration

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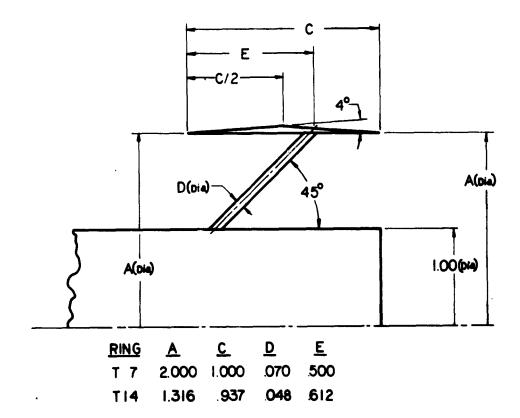


RING	A	B	Ç	D	Ε
TI	1.500	1.605	.750	.046	.375
T 2	1.500	1.657	1.125	.052	.563
Т 3	2.000	2.210	1.500	.080	.750
T 4	2.000	2.140	1.000	.070	.500
T 5	2.000	2.080	.600	.060	.300
T 10	2.500	2.675	1.250	.092	.625
T 11	2.500	2.605	.750	.080	.375
T 12	1.250	1.425	1.250	.048	782
T 13	1.250	1.381	.937	.048	.623

## (a) 4° INTERNAL EXPANSION

#### ALL DIMENSIONS IN CALIBERS

Figure 2. Ring Configurations (All Dimensions in Calibers)





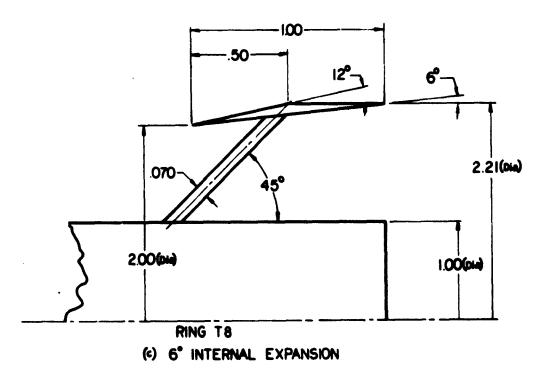
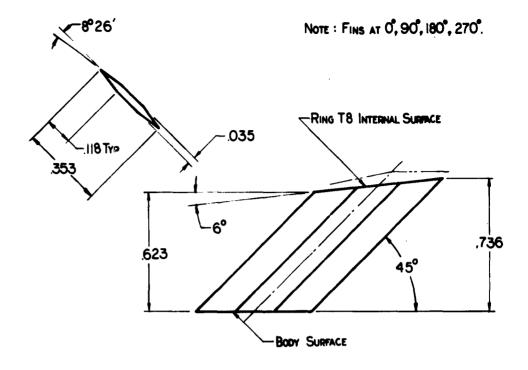


Figure 2. (Concluded)



FIN NO. 8S

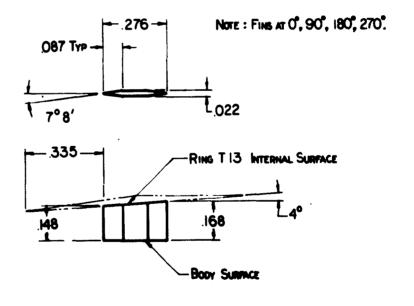


Figure 3. Support Fin Configurations (All Dimensions in Calibers) 7

M FIN NO. 13S

Table 1. Wind Tunnel Test Program

				-	AEDC								BRL			
Config.	0.80	0.90	0.95	1.00	1.05	1.10	1.20	1.30	1.50	1. 75	2.00	2.50	3.00	3.50	4.00	4. 50
Bl B2	2, 3, 5	2, 3, 5 5	2 5	2, 3, 5 5	2 5	2,3,5 5	2	2,3,5 5	2,3,5 5	1,4	1,4	1,4	1,4 1		1	1
BITIXI X2 X3	2 2 2	2 2 2	2 2 2	2 2 2	2 2 2	2	2 2 2	2 2 2	2 2 2	1 1 1	1 1 1 1	1 1 1	1 1 1		1 1 1	1 1 1
B1T2X1	3	3	3	3	3	3		3	3	1	1	1	1		1	1
X2 X3	3	3	3	3	3	3		3	3	1	1	1 1	1		1	i
B1T3X1 X2 X3	5 5 5	5 5 5	5 5 5	5 5 5	5 5 5	5 5 5		5 5 5	5 5	1 1 1	1 1 1	1 1 1	1 1 1		1 1 1	1 1
BIT4X1	5	5	5	5	5	5		5	5	1	1	1	1		1	1 1
X2 X3 X4	2 2	2 2 2	2 2 2	2 2 2	2 2	2 2 2	2 2 .	5 5 5	5 5 5	1 1 4	1	1 1 4	1 1 4		1	i
B2T4XI X2	5	5	5	5 5 5	5 5 5	5 5 5		5	5 5 5	1 1 1	1 1 1	1 1 1	1 1 1		1 1 1	1 1 1
X3 B1T5X1 X2	5 5 2	5	5 5 2	5 2	5	5 2	2	5	5	1 1	1 1	1 1	1 1		1 1	1 1
X3 B1T7X1	2	2	2	5	5	2	2	5	5	1	1	1	1		1	1
X2 X3 X4	5 2 2	5 2 2	5 2 2	5 2 2	5 2 2	5 2 2	2 2	5	5	1 4	1	1 1 4	1 1 4		1	1
B1T8X1 X2 X3										1 1 1	1 1 1 1	1 1 1	1 1 1		1 1 1	1 1 1
BIT8SXI X2										1 1 1	1 1 1	1 1	1 1		1 1 1	1 1 1
X3 BITIOXI X2 X3						-				1 1 1	1 1 1	1 1	1 1 1		1 1 1	1 1
BITIIXI X2 X3										1 1	1	1 1 1	1 1 1		1 1 1	1 1 1
BITIZXI X2 X5	5 3 5	5	5	5 3 5	5 3 5	5 3 5		5 3 5	5 3 5	:	1	4	*			
B1T13X1 X2	3	3 3	3 3 3	3	3 3	3 3 3		3 3 3	3 3	4	4	4	4 4			
X5 BT13SX1 X2	3	1 3	3	3	3	3		3 3	3 3	4	:	4	4			
X5 BITI3RX	3 3	3	3	3	3	3		5	5	4	•	4	1			
	23 3	3	3	3	3	3		3	3			4	4			
B1T14X1 X2 X5	3	3 3	3	3 3 3	3 3	3	-	3 3	3	4	4	4	4			

#### TEST NUMBERS:

- BRL-1838; 13-23 May 1963
   AEDC-TM173; 19-22 November 1963
   AEDC-TM193, 1 11 September 1964

- 4. BRL-901; 1-11 September 1964
  5. AEDC-TM193; 21 September 6 October 1964

Table 2. Index of Tabulated Data

Table Number	Config. Number	Table Number	Config. Number	Table Number	Config. Number
3	Bl	21	B1T5X1	39	BlT11X3
4	B2	22	B1T5X2	40	B1T12X1
5	BITIXI	23	B1T5X3	41	B1T12X2
6	B1T1X2	24	BlT7X1	42	BlT12X5
7	B1T1X3	25	B1T7X2	43	B1T13X1
8	B1T2X1	26	B1T7X3	44	B1T13X2
9	B1T2X2	27	BlT7X4	45	B1T13X5
10	B1T2X3	28	B1T8X1	46	BlTl3RX1
11	BlT3X1	29	B1T8X2	47	B1T13RX2
12	B1T3X2	30	B1T8X3	48	BlTl3RX5
13	B1T3X3	31	B1T8SX1	49	B1T13SX1
14	B1T4X1	32	B1T8SX2	50	B1T13SX2
15	B1T4X2	33	B1T8 <b>S</b> X3	51	B1T13SX5
16	BlT4X3	34	BITIOXI	52	BlTl4Xl
17	B1T4X4	35	B1T10X2	53	B1T14X2
18	B2T4X1	36	B1T10X3	54	B1T14X5
19	B2T4X2	37	BITIIXI		
20	B2T4X3	38	B1T11X2		

Table 3. Aerodynamic Coefficients for Configuration Bl

.4190	CAF	.117 .115 .113 .108 .105 .115 .106 .107	.4160	CAF	971. 184. 179. 171. 771.	771. 771. 169 1631. 271.	.3200	CAF	.145	150	147	151				
N. N	CAB	.220 .210 .205 .205 .204 .204 .207 .219 .219 .235	R 14	CAB	189	.188 .198 .212 .245 .183	S S	CAB	.135	127	123	.125 .127 .127 .127			•	
7.090	ž	.603 .439 .273 .125 .125 .229 .229 .229 .1174	8.400	£	.349 .327 .227 .112		4.930	5	1.145	249	021	. 225 . 225 . 298 . 465				
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0607	CAF	119 1120 1120 1120 1120 1120 1120 1120 1	.4320	CAF	.181 .187 .186 .175	.174 .175 .169 .156	.3400	CAF	.157	158	159	161	.326.	CAF	.118 .118 .118 .118 .118 .135 .157 .157	
z z	CAB	.178 .169 .158 .157 .158 .162 .170 .187 .187	N K	CAB	.207 .196 .189 .195	. 188 . 199 . 207 . 225 . 267	N N	CAB	161	142	143	152	α Z	CAB	0.050 0.050 0.050 0.050 0.050 0.050 0.050 0.050 0.050 0.050 0.050	
6.750	ž	. 551 . 394 . 251 . 089 060 369 545 545 545	8.190	5	.597 .457 .324 .195	087 226 371 552 952	5.850	5	930	212		.174 .232 .364	2.450	3	-1.859 -1.329 -1.329 644 371 187 058 -173 173 753	
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Table 3. (Continued)

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# Table 3. (Concluded)

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*	8				1136 1136 1136 1136 1136 1136 1136 1136
. 6.520	5		1.540		. 250 . 200 . 000 . 000 . 200 . 243 . 243 . 257 . 100 . 100
9	5	11.1 6000 71.000 71.000 71.000 71.000 71.000 71.000 71.000 71.000	95.	3	11132 10133 10133 1013 1013 1013 1013 10
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Table 4. Aerodynamic Coefficients for Configuration B2

*190	CAF	140	.153	.154	152	147	142	157	į	.3.63		4	.176	1.45	.147	197	.176	.172	166	.182		.3100	CA F	.143	5.5	.151	941	.152	7	200	141		.145	.152													
z	CAB	950	.054	150.	250	660	190		l l	* *	: ;	0 4 9	*11*	13	001.	101	110	-117	143	101		z o	CAB	690*	990	590	4	.063	290	.062	-062	60.	•064	• 0 6 5													
7.090	5		062							8.560		5	.321	.143	280*	120.	112	149	705	•025		- 4.130	5	-1.225	822	385	220	089	032	. 66.	95.7.	283	470	*69.													
8	Š	721	050	022	50	• • • • • • • • • • • • • • • • • • • •	.153	92.	ŝ	5	2	Z U	168	119	032	10.	660	.145	199	.0		3.00	3	.379	-287	155	160	0.0	•10.	210	067	.00	2	246													
H 1.00	ALP		-3.18							64.5		ALP	-4.35	-5.25	-1.06	60.	2.22	3.32	4.4	60.		¥ .	ALP	26.6	4.91	3.89	1.83	1.29	-26	- 25	-1.27	1.1.	-3.29	-4.32							•						
06.07	CAF		133	136	136	133	.131	107	.136		.4160	CAF	.182	186	187	.165	.182	25.	.168	.165		3260	CAF	. 157	154	.155	941.	.155	150	.154	151	152	.155	251													
	8		450								· · · · · · · · · · · · · · · · · · ·	843								940.			CAB		082	970	520	720	*	50.	2.0	.075	9.5														
6.750	5		121.								004.	5								598		4.930	5											245													
- 0 56.	8		126								1.30 0 .1	3								305		2.50 0 =	ē									_		150													
		•	-4.23	-2.10	9	2.15	7.5		.02		*	ALP		-3.22	-2-13	1	1.13	2.22	1.4.4	9.6	5	=	•	•	00.0	3.95	×.	<u>*</u>		-	£:	-1.75	-2.27	-3.27													
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	•		126								1.10 0 -	2	5	153	.07	•034	8 6	.0	61:	275	ş.	8		5	96.		2.5	200	9	8	520°-	070	-00	-139	.50	3	,	24.5	2	2 5	20.	9.0	410	.075	7	202	
	-	ALP	4.22	-2.10	4 % -	1.00	3.20	4.74	0.00			3	ì	4-20	-2.13	-1.05	ç 5	2.17	3.25	6.3	*0*			4	41.0	4.09	3.04	70.7	•	-0.	34.	7.1	47.2-	-3.15		•	ł	8.6	2.5	1.7	17-1	.21	?:	·			
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92.	3	5	-11.	6700	028	.078	129	-196	0	160	7.40	;	5	797		160	250	770	- 045	• • • • • • • • • • • • • • • • • • •	.0.7	٠,٥٠	;	S	712		062	10	055		20	0.	-	* · ·	36.7		5	-1.500	14.	90	*	120	.09	2	Į.	7.4	•
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Table 5. Aerodynamic Coefficients for Configuration BITIXI

M = 1.00 Q = 7.090 AM = .4190	ALP CN CM CAR CAF	-4.35376 2.245 .193 .473 -3.25261 1.545 .185 .483	-155 .065 .180	.071 609 176	.177 -1.295 .1 <sup>8</sup> 0 .284 -1.968 .185	.391 -2.639 .194	732 -4-798 -201	.071 616 170	C-11-30 3 # 6.460 3N# 4163	CN CN CAB	360 2.061	248 1.386 .187	039 .137 .184	.066490 .184	.274 -1.722 .162	.375 -2.309 .183	6.77 .705 -4.208 .213 .402	.064462 .193	M = 2.50 G = 4.930 RM = .3200	ALP CN CM CAB CAF	.496 -2,378 .135	.397 -1.893 .134	222 -1.061 -132	.000 465 -132	.056266 .132 .015076 .133	023 .113 .133	-103 .494 .133	144 -492 -132 185 -491 -132	-3-37265 1-276 -131 -276 -4-40350 1-684 -131 -271												
0904. * NF 057.6 # 0 49. * M	CAB	-4.32355 2.091 .139 .480 -3.24251 1.449 .139 .485	240143	.050432 .137 .157 -1.008 .138	.251 -1.667 .135	141. 476.2- 206.	715 -4.642 .146	.656463 .139	. 8.190 AM	ALP CN CH CAB CAF	1917 1-056	231 1.247 .179	139 .720 .175	.041270	.132171 .166	.320 -1.847 .179	6.71 .648 .3.783 .207 .458	.039256 .170	H = 2.00 G = 5.850 RM + .3400	ALP CN CN CAS CAF	.491 -2.447 .165	.309 -1.520 .157	.223 -1.093 .156	200	.057262 .136	021 . 104 . 156	-100 -403	-142 .700 .135	-3.37362 1.302 .153 .298 -4.41348 1.721 .153 .298	M = 4.50 G = 2.450 RN = .3200	ALP CN CN CAS CAF	5.60 .529 -2.692 .057 .230 4.61 .417 -2.066 .057 .236	.313 -1.522 .057	750° 479° 971°	. 062422 .057	750. 600 000.	062 .401 .057	- 124 .405 .057	150. 600.1 502	347 1.894 .057	
M = .90 G = 6.520 RN = .4010	ALP CN CN CAB, CAF	-4.29319 1.786 .116 .397 -3.21215 1.131 .108 .401	036 .087 .100		.230 -1.423 .111	.330 -2.061 .115	.703 -4-509	.055 437 .100	H = 1.10 G = 7.700 AM = .4230	ALP CN CH CAB CAF	321 1.767 .187	219 1-170 -181	041 .154 .169	.132336 .163	.215 -1.274 .167	.314 -1.856 .182	6.65 .651 -3.612 .215 .482	.042316 .160	M + 1.75 G + 6.200 RM3600	ALP CN CN CAB CAF	.526 -2.783 .171	.324 -1.689 .162	1228 -1.176 .161	.003076 .155	.010051 .151	020 .152 .151	114 .578 .194	261 1.026 .160	-3.44284 1.476 .161 .332 -4.51361 1.462 .161 .337	n + 4.00 0 = 2.340 4m = .2500	ALP CN CM CAB CAF	5.71 .534 -2.691 .070 .216 4.71 .415 -2.626 .870 .207	0.00 0.00.	. 137 -1.026 .076 .	.000436 .071	.000035		. 140 . 140 . U.S.	157	261 1.316 .073 .	
Clet Mr 007.6 - 0 08 4	3	-4.27320 1.752 -139 -396 -3.19212 1.081 -333 -403	111 -473 -130	.063525 .126	.244 -1.574 .134	.353 -2.296 .130		.041 517		M . 1.05 G . 7.460 RW4220	ALP CN CM · CAS CAF	335 1.904 .198	226 1.231 .184	036005 .170	.056448 .170	401· 44·1- 44·	.340 -2.148 .394	6.5			H - 1.50 G = 8.560 4M = .3860	ALP CN CM CAB CAF	361 2.017	VOIC		00+ 181 + 400- 141 + 41-1	.260 -1.971 .101	191 - 2-567 - 187	181. 485 240.	COLE MA 061.4 - 0 00.6 - H	ALP CN CN CAS CAF	-2.419	.300 -1.369 .107	.210997 -100 .127580 .107	101.	.005023 .106		101. 155. 611		101 1.272 107	

Table 6. Aerodynamic Coefficients for Configuration BITIX2

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7.090	5	2.054 1.359 1.359 2.035 2.252 2.526 2.526 2.536	995° E	2.478 1.769 1.057 .300 394 -1.117 -1.799 -2.514 -3.814 -3.853	£1.4	2.5.1.1.2.2.4.2.4.2.4.2.4.2.4.2.4.2.4.2.4.2
•	3		•		• •	n,
8.1	ŭ	- 229 - 229 - 121 - 024 - 075 - 170 - 372 - 474 - 474 - 404	1.50		8.8	23.43 2.343
	414	-3-30 -2-3-22 -2-13 -1-05 17-12 2-21 2-21 2-21 2-21 2-21 2-21 2-2	, ¥	1.007 1.007	. 3	**************************************
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<b>9</b>	CAF	44444444444444444444444444444444444444	.4160 CAF	1000 1000 1000 1000 1000 1000 1000 100	3200	48444444444444444444444444444444444444
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Z	3	222 221 221 220 203 203 203 222 222 222 222 222 222	2 3	233 224 225 225 226 227 227 227 227 227	* 3	771777777
6.750	*	1.226 1.226 1.226 2.276 1.635	} 5	2.748 1.924 1.924 299 349 3.14 3.184 5.184	Š 5	2
•					• .	•
.95	Š	2115	30.1	- 294 - 294 - 176 - 055 - 055 - 055 - 180 - 537 - 537 - 780	2.50	25.5.5.5.5.5.5.5.5.5.5.5.5.5.5.5.5.5.5.
	ALP	-5.28 -7.28 -1.04 -1.04 -1.03 -1.03 -2.19 -2.19	· 4	4.4-0 1.05-1.05-1.05-1.05-1.05-1.05-1.05-1.05-	<u> </u>	**************************************
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010	CAF	1000 1000 1000 1000 1000 1000 1000 100	320 CAF	71111111111	66 40 CAP	2247 2300 2300 2300 2300 2300 2300 2300 230
3	J		. 4320	* *********	.3400	•
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6.520	5	2222222222	\$ 5 5	\$0.55 \$0.55 \$0.55 \$	5.850 CH	2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2
		2.245 1.595 1.744 		2.639 1.601 1.008 1.361 1.008	•	44441111111
9	5	252 - 1342 - 1343 - 134 - 135 - 135	. 02·1	1172 	8	6. 5. 5. 5. 5. 5. 5. 5. 5. 5. 5. 5. 5. 5.
	4,6	1.04 1.04 1.04 1.04 1.04 1.04 1.04 1.04			. ş	***************************************
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.3810	5	417 420 420 420 439 410 410	₹. 5	************	.3400	25. 25. 25. 25. 25. 25. 25. 25. 25. 25.
£	3	191 189 189 188 188 188 188 188 188 189 189	# # W	2 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	. 5	# # # # # # # # # # # # # # # # # # #
\$.700			÷ .		8 s	9
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?			6.7		2.1	
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Table 7. Aerodynamic Coefficients for Configuration BlTlX3

.4190	CAF	4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4	0414	.367	555	2, 5, 5	96	.3200	4	\$ <del>\$</del> \$ \$	207	.291 .294	20.5	262.298								
z z	843	38 1 2 2 3 2 3 2 3 2 3 2 3 2 3 3 4 9 3 3 4 3 3 4 3 4	295 295 RN =	. 286 286	252.	25.23	<b>%</b>	2 .	<b>5</b>	200		133	132	133								
7.090	Ŧ.		8	£ 7	\$ 0. 9. S	28.85	2	4.930														
•	2	3 1.700 1.093 1.093 0.062			1.364 9 .729 3 .136																	
1.00	U	114111111111111111111111111111111111111	3.30	332	677-	1.45	80.	2.50						- 123 - 167 - 212 - 301 - 301								
¥	ALP	-4.29 -1.05	E .	46.95	1.05	1 4 4 9	\$	*	A C			¥.		7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7								
0600	CAF	2000 2000 2000 2000 2000 2000 2000 200	.363	CAF.	4 9 7	.466	\$ 7.7. 2.7.7.	.3400	CAF		315	316			.3200	CAR	22.51	1 6 T	22.	7.5	- 8 £	102.
2	5	24 24 24 24 24 24 24 24 24 24 32 32 32 32	233 24 =	316	25.5	259 269 286	.313 .359 .254	£	<b>C</b>	225	225		. Q	22.22.2	ž	3	\$\$\$	3 5	; ; ;	\$ \$		•
4.790	5	1.872 1.201 .639 .117 944 -1.476 -2.068 -2.76	6.190	÷ 6	285	3.56	.536 .958 .272	5.850	5	¥5.5	***		E.F	1.043 1.943 1.912 2.472	2.450	5	-3.013 -2.333 -1.726		145	5.5 5.5 5.5 5.5 5.5 5.5 5.5 5.5 5.5 5.5	1965	8
•	3	-311 -120 -120 -037 -044 -135 -135 -135 -135 -135 -135 -135 -135	•					•						1125	9	5	. 550 . 324 . 124					
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<b>4010</b>	CAF	.280 .280 .826 .826 .826 .296 .296	32.	14.14.		457	÷÷.	.3400	CAR	.335	955		¥.		.2500	CAR	24.1	100	==	:::		
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6.520	5	2.005 1.292 .711 .147 935 935 286	7.700	<u>د</u> و د	102	ESE	355	• 200	5		13		÷ ?	******	2.340	5	2.994	:: <b>;</b>	29	ij	ê eş	<u>.</u>
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<b>?</b>			<i>:</i> :		B 21			1.75							8,4	Ī	žė,					
*	7	12.12.12.12.12.12.12.12.12.12.12.12.12.1	Ö ,	4.32	-2.15	2.18	4.4 6.6	*	3	***	~ ~ .		~ ~	7777	*	1	***	138	2.8	;	777	
0101.	443		.22.	300			Ç		.3860	378			* .	26.8	.3100	Caf	52: 23: 86: 86: 86: 86: 86: 86: 86: 86: 86: 86		22.	.235	87. 22. 26.	.23
:	8	22.00 22.00 22.00 22.00 22.00 22.00 22.00 22.00 22.00 23.00 23.00 23.00 24.00 25.00		5 %	, , , , , , , , , , , , , , , , , , ,		70 70 70		:	CA8		221	7.5	222	*	5	466	22		Ę	601.0	701
5.100	₹.						3.739		0.50						4.130	8						
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Table 8. Aerodynamic Coefficients for Configuration BIT2X1

8	CAF	549.	8	.653	9	3	999	. 657	949	. 629	70		3860	A P		1	3	1		1	465	1	*		.3100	7	į	-274	2.5	2.0	. 291		207		.293	2	Š	.276	2														
Z	8	662	96	4	2		6	9	• <b>2</b> 0	573	*	2	3	84.		**	3	791		1 3	3	991	•	3			9	601	607	601	2	25	2 =	2	Ξ:	22	2	6	5														
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.00	5	3.622	2.124	1.255	7		-1.679	-2.517	-3.349	-5-115	747	-:13	. B.560	Š	•	3.096	9	.629			-2.71	-3.554	-5.		; ;		;	-2.96	-2.335	62	0			=	3		0	*	7.0 7.0														
·	3												•	Ž		***									•	2	5	.575	456	256	-159	01:	510	.031	920	771	215	700	Ş														
8	ALP												1.50	41.6											3,00																												
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8	CAF	6	4	27	ş	~:		. 5	2	87	92	8	9919	. ;	4	93	9 :	3.5	35	5.	27		4	\$	.3200		Š	323	2	220	358	335	25	25	324	322	926 926	327	326														
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8.7	5	7.114	.267	189.	2	2	2	704	,	.397	.13	.319	9	3	5	.033	9:	262	•15	:			404	.145	4.9 X		5	1.126	\$				ž		322	\$		575	:::														
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.40	Q.	8	\$	.5	3				. 531	.53	. 52	.52	.4230		5	9	•	5	ě	•		\$			946	•	5	,			.3	•			÷		•		376.	.3200	3	52.	ž.	ç	ž	2:	??	.2	21	*	ž.		!
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6.520	5	8	•	7			2	8	\$	;	2	2	7.700		5			2	11	7	: 3		5	=;	9		5		21	332	3	2 9	:	2	221	í	ī	326	53	3.490	5	*	<u> </u>	51	<u> </u>	9	**	ž	Zì	i£	2	î	į
•	5																							-1.081			_												2.495						24								
9	5	470	319	192	2		278	*	.587			Š	1.10		5	513	900	107	.02	.169					8	3	5		7	ğ	Š	<u>.</u>		ô	200				327	8	5	ž	į		<u> </u>	=	.02	.0.	3 9	. 152	Ž		ĺ
	ł	.34	3.24	<b>*</b> :	8	5	-	5	7	29.	715	ē.		:	ì	7.	۲.	6	60.	*	5.5		22.	1:0:	•	٠	414	, '	6.6	8	\$.2	2	:	7	?		.75	2.20			1	9.90	2		!!	<u></u>	1	:	<u> </u>	2	2.5	2 2 4	}
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.3816	CAF	413	. 337	ž	**	. 559	22	, .	536	33	. 554	. 551	0224	C.A.F.	į	7.		\$	669		.69	•	1		9		CAF	:		.422	7		,	î	.432			?	??	->200	<b>4</b> 0	.25						. 220	. 23.	2	6		
	5		131	52	221	=	- :	2	5.5	9	119	171			}	250	2	96.	e ;	ž	21.	221	236				640	;	7.5	192	2	2 2	! !	₹	2			ĭ	 		3	0.73	220	57.0	27.0	56	50	2	520	079	2.5	! !	
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\$ .700	3	2.33		8	*	0		77	-7.51	-4.170	8	. *27	7.5	5	,	3.32	7.00		7			-3.63	-5.59	*	•	;	5		7	-2.67	1			~			-	1	2.142	÷	5	-3.10	7.7	2		*		9		2	81		
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•	•												1.05	•													•														;												
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Table 9. Aerodynamic Coefficients for Configuration B1T2X2

÷.	CA	3	3	3 3	3	3	8		. 546	3.	3			9	CAR		.485	4 4 6		8	8	9	1	1	•	64.		.310	3		787	296	Š	.29	. 29	295	-287	.302	962	7.65	2	.290															
Z	CAB	.253	7.	2		747	.251	797	,266	<b>584</b>	245			Z	CA8		518	219	222	220	250	218	917		221	.221		*	CAB		9116		118	27:	27.		123	123	22:	**	*21	123															
7.090	<b>.</b>												;	2	*			•			Ī							8.	*																												
			1.535											•	_										1			•			-3.352																			٠			-				
00.1	3	175		*			Š	324	450	.755	000			Ş.	5	,	545	382	0.0	0.	. 202	¥.		:	1	4		8	3		404	373	.268	17:	90	.021	026	075	9	222	317	423															
-	416	17.40	-3.22	-2.13	-1.09	=			4.30	9	.02	}			ALP	Ì	14.4	-3.32	-1-07	ð	1.1	2.31			-1.01	•0•	٠		ALP		5.91	3.87	2.03	1.50	2.7	.23	27	9:	2011	-2.34	-3.33	#. 1															
2	Ç.	.556	. 559	766	. 553	.550	**	.534	.536	.545	. 542			.4160		CAF	773	į,	.549	.550	166.	7	530	.527	. 525	.543		3200	, TAE	į	-315	121	.331	.328	.327	323	330	. 337	.327	327	335	.314															
	84	218	-214	ê	28	2	218	228	238	239	0K.2					9	***	227	223	727	*22	• • • •	227	235	. 255	227		:			7:																										
R	*													9		5												98																													
	_		1.754																						0.0			•			-3.704																										
÷	5	30	273		0.02	.123	.227	ž	.482	.762	.342			1.30		3	. 613	.37	245	2	6		454	.603	:	. 360		2.50	ā	,	-632		.20	9	2	9	-0	\$	===		-	430															
•	46	-4.31	-3.22	-2-1	60.	1.11	2.19	3.28	4.37	15.9	3.28					46	1	-3.31	-2.20	-1.07	ξ <u>-</u>	2.2	3.40	4.53	•••	-3.31		~ .	414	i	<b>6.03</b>	3.6	2.93	1.91	1.37		7	o	-1.21	-1-1-	-3.25	-4.20															
:	C. B. F.	.493	\$ <b>\$</b>		•	<b>‡</b>	٠.	.48	*	.497	į	*		04.24	2	CAF				565		.650	•		0	.65	260.	.3400	•	į	305	200	.363	.300	.36	387	.307	.379	*	796	.386	.376	.3200		•	242		***	.234	.240		.234	192.	.231	240	330	
;	5	.163	151		137	₹.	÷	.155	3.	20.	51.	. 1 52		2	•	3		*2.		253	*52	3.	į	į	200	Ŕ		=	5	į	27.	1	3	.162	=		3	:	=	2	791	<u>:</u>	:	;	5	960	560	6	0.00	055	200	200	550	500	6.00	200	
	5		2.43											7.300	3	5												. 5.050		į	162.4-												2.450		5	-3.40											
; ?	3		¥.											•	•	3						•				2		•	3	;	9												•														
•	41.6		-3.23													AF.												M . 2.00	•	ì	£.03												# 4.50			9.6											
	*	186.	.533	, , ,	.538	. 537	.535	. 523	. 520	. 515	ī.	. 531		0224		Ç <b>V</b>	;		119.		7		1	1	044		•	.3000		<u>;</u>	704.		9	• 0•	7	-		•1•.	=		\$ 5	<b>:</b>	.2500	•	į	**	-209	266	245	.267	157	562.	-240	267	9	7.	
	<b>64</b> 0	185	:			172	170	90	.105	00.1	717	:				<b>8</b>		ì	\$	200	1	<b>2</b> ;		9	£	2.			:	į	.10		=	98.	2		-	111	2	2!		701			3	\$ 0	\$		2	5	ė c	2 2	2	5	2 2	2	
8	5		20.2											7.40		5										2.003		• 200		5	******												2.340	;	5												
•	3		~					212	9.6		- 1112	ş		•	•	5												•		Š	.735												•	· :	5												
₹.	٩													1.05		1										2.		1.73			•.09												9	•	}										7		
*	•	1	Ť	7	~	•	•	• •	• •	•	•	•		•				11	7	7		-	~ ′	•	•	7 1	7	•			•	•	~ ~	-	-		•	•	7	71	``	*	•			•	•	_ •	• -			•	•	7	77	7 7	•

Table 10. Aerodynamic Coefficients for Configuration B1T2X3

•4190	CAF	.550	.556	55.	.563	266		545	126.	. iS	3		Š	***		2.	. 469	*	į	\$	8	164.	<b>~</b>		.3100	4	ב ב	.303	Š	298	30	. 292	2 6	304	. 292	-292	.301	. 30														
ž	CAB	318	313	2	967	2	323	.335	336	, K			CAB		3	.231	.223	250	577	236	.251		.218		z	;	9	0110	2:	911	901	107	95	102	901.	90.	801	601	:													
7.090	5												5												.130	;	5	1 64	123	606	7 7 7	90	<b>.</b>	502	551	876	213 951	2.189	2													
	_									1.265													. 183				5																									
9.1		614	.30	073	.03		37		=	180	:	2	ō			52		6		Ť	9	•	0.0		3.00		ü											-354														
	A L P	4.31	-3.22	-1.05	6	= :	3.28	4.36	6.55	-2.13			4	•		-2.20	-1.00	8	::	3.30	4.51	9.	:6		*		4	5.8	4.0		1.78	1.22	£:	3.5		-1.33	-1.65		•													
£	ų.	2 3	2	-	2 2	2		. 1	10		8	ų		5:	2 5	2 2		2.	<b>:</b> :	. :	. <del></del>	•		2	ų		2:	2 52	*	2:	2 =	: 1	2	::	. ~	2		•														
. 40 %	CAR										•		į											• .3200	3												345															
2	3	. 281	.252		.251	.261	2,0	316.	.261		*	3	,	-266		.253	. 253	.256	25	273	8	.252		*	20		541.		-	3		*	. 145	-	-	.147	::	?														
6.1%	5	2.440		8	685	1.35	2.093	4.002	1.352		8.400		5	3.776		704	319	1.331	2.362		6.031	.695		¥.9	ŭ	•	. 29	2.778	2.017	262	27	225		7		1.520	2.23															
•	3	365									•	3		511										•	N																											
.95	41.	٠									. 1.30	•												. 2.50	4												2.5															
•	•	-4.28	?	÷	• -	~	ď.		~	i	•	•	•	-4.40	•	;	: •	-	~•	•	•	÷		=	•		•		~	<u>.</u>	'	• •	i	•	7	7	Ť	•														
0104	Z,	£1.5		.437	. 437	.420	614.		.423	724.	.4230	•	5	• 60				.616	+19.	8 5	ŝ	879.	ş -	;	3400	CAF	;		36.	98:		300	.36	2	*	.386	1		3200	4	;	276	1	.262	.2.	.259	.253	.259	.257	.259	.262	!
	9. 5	292.	215	9	202	230	5.5	787	230	217,		;	E 4	319	324		2	3	305		329	330	. 255 255			CAB		2 2	213	220	522	\$25	526	225	226	526	227	727		840	. ;		240	· ·	643	740		7.7	740	140		:
6.520	*										7.700	;	5												200	=													2.4%													
•	_	3.423									1 - 0												-1.63		•	Ü			-3.1	-2-3			× .	ē 3	•	1.3		3.33	2.	U												
\$	5	7	- 172	90	.121	.242	.3	7	.24	- 10	1.10	į	5	624		9 9	0	.020	.127		•	.757	200		00.	Š	-	200	.462	<b>1</b>	522	102	Ş	.013	- 13	-192		•	• 50	5	. :		365		7115	0	02	- 07	:	217		
•	ALP	-4.2	-2.12	-1.05	92	2.17	3.25		5.16	-2115		:	ì	-4.34	1		90.1-	70,	1:1	7.7		19.9	2.5 2.6	,		414	:		•00	3.02		•	.3	7.	:	1.01	2.5		•	A.C.			3.78	2.70	1.21	27.	: °	71.		2.35		:
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. 3810	3	. 457		9,	1		1	•				:	3	598	ŝ	1		. 612	004	66.	ŝ	8	<b>÷</b>	. 3600	;	•	Š.	?		47.	2		ů.	*		?		•	.2500	5	.242	3,50		2		. 23			. 236		**	
2	C A 8	.273	1:	127	.23		7.			6	ţ	:	5	341	*				1328			.324	. \$22	*	•	•	.263		275		-712	7.00	797	?	200		545	<b>.</b>	=	8	700	290	700	200	7		600		.0.	.00	.003	
\$.70	5		<u> </u>	?	4	,	*	3	? ?	3	1.000	;	<b>.</b>	0.0	20.	:		411	\$00	53	ì =	-1.002	36	002.4	;		352	;;	ĝ	<b>19</b>	767		711	578			2.71	2.		5	•	•	: 5		£į	. 8	0.1	700		::	ĩ	
9	5										3		•											•	,														•													
÷	~			3							.00											. 23		1.73	•												-		8.	ū												
×	•	****	-	0	7		?	. 30					4		7.1		ò		7.20			1.19	•1.7-	:	•	•	.17	• • •			?	: :	15	:		07.7	• • • •	?	*	4	0	~	3.74		77		2	7.7	0	. 3.91	7.3	

Table 11. Aerodynamic Coefficients for Configuration B1T3X1

•	J	33	3	-	2	3	3	2	3	2	:		7	J	•	•	المر	ļ			•			•		•		٠	•	٠	• •	•	•	• •	•	•	• •	•													
*	CAB	.237	.217	į:	777	2	-222	2.5	3		167:			CAR	į	f	1	•	•	1,4	ξ.	1360	*254	*77	i	Z	5	.115		911:	:	571	27.	2 2	2.	2	22	22													
1.090	5	8.745	404	1.682	0.0	-5.615	-7.895	100	14.311	ŝ	121-01						*	212		-2.875	-5.111		-14.364	200		•	ž	901-4-	-5-137	-4.011	-2.940	1.301	790		.774	1.207	2.350	3.333													
8:	3	1-144	200	2	6	727	1.029	1.312	1.092	200	1.328 -		1		Ś	4	. 65	Ž,	45.20	4	.670	1.299	1.937		;	9	3	470		9.			120	600	-	-19	352	215													
	ALP	- 4.70	-2.33	*1:7-	ę.	19.7		÷.	61.2	ŧ ŧ	4.82		2.		317			7	-1.1-	1.2	2.52	5.00	7.51	7.51		i E	**	4.03	16.4	3.95	2.90	1.30	•	.26	:	-1.28	-2.33	-3.32													
0,040	CAF	940	1.059	1.077	1.001	0.00	1.038	1.034	1.028		.4100		CAF	.796		-832	940	848	100	.623	.743	.658		•3700	CAF	125.	530		.543	514	. 536	.530	. 531	.534	-516	. 5		٠.				,		~			+	vite,		٠.	
e R	<b>8 V</b> 3	3:	25	153		15	3	.153	3:	Ŗ	2		845	400		517		160	•	202	6	184	1	2	45	155	.155	2	.157	3.		.176	2.5	176	51.	2 3															
2	5	1.398									9		5		7.00.7								•	Š.	5			3.311																							
									,				,							•	•			•																											
÷	5	085	745	- 20	2	9	2.	1.03	1.67		5	?	ప	-1.25	616	- 56	77	*	2	2 4	26	Č.	:	2.50	5	3	ě	1	. 32	Ç.	20	05		10	*	7.															
*	*	8° 4	-2.27	-1.12	.09	1.10	05.	3	.0	19:6-	•		ALP	0,4-	-3.65	-2.40	62-1-	1.32	2.57		6.19	.01			414	<b>♦.1</b>	5,12	3.00	1.97	3.6	4 15	13	100	2	-2.25	2,5															
0104	543		7	2				516	. 40	٧. ٠٠	04.2.30	!	3	\$2.001	980		1,053	.055	500	14043	.C28	.056		9460	CAR	000	. 542	0 to 10 to 1	362	060	400	0 40	586	2	200	585	***	;	3700	CAF	.3%	\$		0	.38	.392	38		.300		
	8	110	ē		\$	3		707	135	106			<b>9</b>	£ .	Č,	197	747	2	200	8	752		!	4 2	673	**		: :	J	7.5	100		•	156	1.51	181	156	;		( <b>V 0</b>	0.0	050	<b>5</b> 5	3	•••	\$ \$		\$ 5	<b>:</b>	\$	\$ 0
6.520	5		3								7.700		<b>7</b> ,					-				7.000		2	5			-5.05 C											2.430	5	•					738					
•	5	\$	į	į	3	2	~			ž	•		5	327 14	56		8		T 1	- 59	JAN -	2-080°Z	ļ	•	Š															3						<b>6</b> 0.0					
•								_			1.10		•	٠										2.80	61.9														•	٥											
•	•	Ť	-3.60	7		-	~	: •	; n	î	•		4	Ť		7	•	1	•	,	8	***		7	•	4		2.10	• :	-	•	•		•	. 1	ņ	Ť		•	•	•	*	PÅ v	•		8.5	<b>'</b> ;	<i>:</i> .	-		•
.3010	3	***	į		?		į	ş	***		***	2	473	0,00	1.003	1.070	900	100	1-347		2400	4		.3603	547	: :		62.9	9.00	į	. 435	:		î	***		1691		.2500	545	2	***	***		-	01	9	į	614.	ē ş	670
:	3	*1.	<b>!</b>		*	<b>.</b>			-13	<u> </u>		į	<b>6</b>	328	.231	27	2	2	797	£ ;	7.	4		2		;		3	3		:	-		*	ë	22	=		=	3	44	1	1	2 4	Ş	•	1	į	1	6	500
\$.700	8,		÷										5									-1.00		9.70		;		****		0.0	1.342	932		1	-				2.363	• 5	.4.758		-3-623	7	1.103	*		ć	200	1.535	
•	5		7.7								•	•	5		ž	ž	í		*			~						•											3	ij						7					
₹.	3		2:									•	*	,	٠.							Ę																	3.3	•						ē					
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Table 12. Aerodynamic Coetiscients for Configuration BIT3X2

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0614 - 400 0 - 7:090 RM	ALP CH. CH CAB CAP	-4.56981 7.817 .271 1.000 -3.56981 7.817 .271 1.004 -3.56968 3.75 .227 1.004 -3.61202 1.664 .277 1.004 -3.61202 1.664 .277 1.004 -3.60202 1.007 -3.60202 1.007 -4.60202 1.007 -4.60202 1.007 -4.60202 1.003 -4.60202 1.003 -4.60202 1.003 -4.60202 1.003	1.50 d = 8.560 AN	CN	-1-132 9-085 -215 818 6-593 -213 516 4-160 -216 210 1-708 -216	.057435 .215 .361 -2.680 .215	2.44 .946 -7.878 .212 .712 .712 .712 .712 .712 .712 .712	M . 3.00 Q . 4.130 RM3100	ALP CN CM CAB CAF	6.06 1.062 -7.455 110 -441 5.06 .875 -4.176 119 -458 4.01 .889 -4.81 107 -447 2.97 .312 -3.450 105 -447 1.94 .335 -2.402 105 -441 1.84 .335 -2.402 105 -441	. 169 -1.072 .063 -453 .101 .057 -101	-200 1.25 102 1.25 103 103 103 103 103 103 103 103 103 103		
4000 ak = .4090	5	10.00 0.00 0.00 0.00 0.00 0.00 0.00 0.0	1.30 Q # 8.400 Ald a .	CN CN CAB	-1.24 10.178 .251 913 7.520 .249 570 4.716 .246	1.447 -11.464 .255	6-71 1.003 -4.761 .250 -4.22 7-05 1.595 -12.319 .254 .621 09 .136 -1.062 .244 .855 1.28 .453 -1.069 .244 .855	1 2.50 G = 4.930 RN = .3200	•	11.102 - 7.064 1196		• • • • • •		
# .9¢ G • 6.520 RN = .4010	ALP CK CM CAS CAF	1.10 (1.10) (1.1	1.10 0 - 7.765 484230	CN CN CAB CAF	-1.193 9.717 .277 1.027678 7.186 .273 1.045556 4.653 .273 1.045240 2.074 .277 1.063	0007 1.000 0273 1.000 0007 1.0100	2.38 - 4.002.70 - 1.000 -	00% - NN 050°C - 0	**********	6-13 1.234 -9-247 -216 -379 5-09 1.010 -7-546 -217 -594 6-00 -501 -5-646 -225 -602 2-00 -501 -5-846 -229 -596 1-99 -246 -229 -596 1-99 -246 -229 -596	.166 -1.261 .241 .999 .006306 .243 .599 031 .223 .245 .597	-130 1.795 .295 .392 .493 .291 .395 .295 .395 .395 .395 .395 .395 .395 .395 .3	N = 4.50 g = 2.450 RN = .3200	2.0
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Aerodynamic Coefficients for Configuration BIT3X3 Table 13.

	1.30 G = 7.096 RM = .4190	940 840 80	<u>;</u>	0.84 B. 340		154 1.544 .250	. 364 - 370 .266	.572 -4.796 .272	3.40 .854 -6.958 .272 .969 4.53 1.107 -6.215 .277 .965	1.454 -13.440 .302	M = 1.50 Q = 8.560 RM = .3660	ALP CN CM CAB CAF	900	635 7-110 -3-	-2.28530 4.521 .341 .743	.046377	***************************************	245 - 4.276 . 342	000 000 (AAC)	1967 990	Calls - was 081.4 - 0 00.6 - M		CN CN CAB	1.109 -8.291 .101 .907 -6.843 .101	.707 -55.361 .100	.330 -2.532 .093	139 -1.059 .097	790, 272, 240,	197 1.050 ,097	-323 2.436	-2.29 -400 8.114 0000 -453 -3.30 -1.11 4.437 .000 472	776 %-858 .095	•											
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. Table 14. Aerodynamic Coefficients for Configuration BlT4X1

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8 - 5.700 RM3010 R9v A -	CAB CAF ALP CR	3,740 .172 .000 -4,50 -4,70 -4,70 .119 4,200 .110	2,395 .142 .716 .2.22359 2.621 .099 1.150 .138 .73209168 .009	427 .134 .737 .04 .075598598598598		-6.300 .105 .722 4.56 1.010 -7.700	-151 .157 .761 6.56 1.057 -6.163	G = 7,460 RM = ,4220 M = 1,10 G = 7,780	CAB CAF ALP CN CH	200 - 200 -	669° 672° 57'6 000° 682° 682° 682°	3.604 .227 .801 .2.29477 3.606 1.724 .221 .8061.12206 1.661	444 - 750 - 40	200-0- 270		-12.279 .789 .782 7.02 1.486 -11.370	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	H CAB CAF	-7.001 .105 .515	-5.610 .170 .516 6.22 .800 -5.040 -4.546 .174 .517 5.10 .741 -4.935	WEST 1960 WILD OCK STILL SECTION OF THE SECTION OF		190 - 461 - 60 - 190 - 1	100 .177 .925 .42 .059	260 177 250 250 250 250 250 250 250 250 250 250	1.74 .177 .927 -1013 -154 1.047	016-1 755- 6-1- 056- 056- 056- 0-0-5	2.00 - 1.00 - 2.		. O OK** . M	CR CAB CAF ALP CR CR	5.00 .075 .4.590 S.06 .750 .4.590	900'M MOS 58'S 100' 100' 100' 100' 100'	-2-150 -090 -3-0 2-01 -3-0 -2-062 -2-062	1.00 090 090 132 1.00 0.212 1.311	955+ 080* 94* 956* 956* 956*	-175 -037 -328 -23 -027167	190° 190° 64° 64° 66° 690° 64° 60° 64° 60° 64° 64° 64° 64° 64° 64° 64° 64° 64° 64	1.31 - 050 .324 - 1.76 - 1.20 1.276	2.46099 .337 -2.31272 1.665 1.96099 .337 -3.30394 2.433	962'd 926"- 16"9-
Ab3010 A90 A -	CS CR CAB CAF ALP CR	7-199 3-748 1-74 - 6065.50672 6-636 -116	-195 2499 142 716 -2.22 -195 2.621 .000 -155 1416 173 -732 -1.00 -143 1405 4003		000 0 000 000 000 000 000 000 000 000	257 -6.360 .145 .722 4.56 1.010 -7.790		1.05 G = 7.460 RM = .4220 M = 1.10 G = 7.700	N CN CAB CAF ALP CN CH	500 1 000 CALL AND	00-0 M21- M4-M- ODI: LAN: MOD-0 454-	465 3-664 -227 -001 -2.2977 3-606216 1.724 -221 -00611.12206 1-661	***** 150° *0° 100° 522° 826° 100° °	505.4- 865. 72.5 Side 885. 487.4- 166.		1599 -12.279 .289 .702 7.02 1-699 -11.570 .695 .691 .695 .695 .695 .695 .695 .695 .695 .695	1.79 G • 6.286 44 . 3400 mm 2.80 D • 5.85	M CH CA8 CAF	210.13 -7.061 -180.4 - 218.		WEBON 1980 WILD DOWN OVER 950 STATE OF		\$60°- \$60° \$60° \$60° \$60° \$60° \$60° \$60° \$60°	650 - 54. 525 - 526 - 510 - 510-	250	296 1.706 1.77 .527	016-01 755- 10-11 056- 056- 056- 066-5 067-	1.01 0.00 0.17 0.20 0.10 0.00 0.00 0.00 0.00 0.00 0.00		o D DS-4 was not	CH CH CH CH CH	5.06 -0-00 -0-00 -0-00 -0-00 -0-00 -0-00 -0-000 -0-000 -0-000 -0-00000 -0-0000 -0-0000 -0-0000 -0-0000 -0-0000 -0-0000 -0-0000 -0-0000	\$60'6" 60°0 50°0 660' 560' 500' 120' 120'	290.5- 200 000 200 000 000 000 000	116.01 - 212. 00.01   26.0 00.00   212. 15.311	640. Cal. 65. 65. 66. 66. 66. 66. 66. 66. 66. 66	-028 -120 -029 -325 -23 -027 -167	000 000 000 000 000 000 000 000 000 00	-213 1.314 .040 .324 -1.78204 1.274204 1.274	-2-31272 1-665 450 2-464 -099 -337230399 2-493 454 3-284 -069 3-4	962-6 926- 16-4-
8 - 5.700 RM3010 R9v A -	CS CR CAB CAF ALP CR	703 5-7647526044.50672 6-636119	-195 2499 142 716 -2.22 -195 2.621 .000 -155 1416 173 -732 -1.00 -143 1405 4003		000 0 000 000 000 000 000 000 000 000	257 -6.360 .145 .722 4.56 1.010 -7.790		G = 7,460 RM = ,4220 M = 1,10 G = 7,780	N CN CAB CAF ALP CN CH	500 1 000 CALL AND	00-0 M21- M4-M- ODI: LAN: MOD-0 454-	465 3-664 -227 -001 -2.2977 3-606216 1.724 -221 -00611.12206 1-661	***** 150° *0° 100° 522° 826° 100° °	505.4- 865. 72.5 Side 885. 487.4- 166.		-12.279 .789 .782 7.02 1.486 -11.370	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	M CH CA8 CAF	210.13 -7.061 -180.4 - 218.	-5.610 .170 .516 6.22 .800 -5.040 -4.546 .174 .517 5.10 .741 -4.935	WEBON 1980 WILD DOWN OVER 950 STATE OF		\$60°- \$60° \$60° \$60° \$60° \$60° \$60° \$60° \$60°	650 - 54. 525 - 526 - 510 - 510-	250	296 1.706 1.77 .527	016-01 755- 10-11 056- 056- 056- 066-5 067-	1.01 0.00 0.10 0.20 0.10 0.00 0.00 0.00		o D DS-4 was not	CR CAB CAF ALP CR CR	5.06 -0-00 -0-00 -0-00 -0-00 -0-00 -0-00 -0-000 -0-000 -0-000 -0-00000 -0-0000 -0-0000 -0-0000 -0-0000 -0-0000 -0-0000 -0-0000 -0-0000	\$60'6" 60°0 50°0 660' 560' 500' 120' 120'	290.5- 200 000 200 000 000 000 000	116.01 - 212. 00.01   26.0 00.00   212. 15.311	640. Cal. 65. 65. 66. 66. 66. 66. 66. 66. 66. 66	-028 -120 -029 -325 -23 -027 -167	000 000 000 000 000 000 000 000 000 00	-213 1.314 .040 .324 -1.78204 1.274204 1.274	2.46099 .337 -2.31272 1.665 1.96099 .337 -3.30394 2.433	962-6 926- 16-4-

Table 15. Aerodynamic Coefficients for Configuration Bl T4X2

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	<b>18</b>	. 782		.73	*.	25.			5.5	41.	741			442	•	~ 6	•04	• • •			.00	**	. 784	•	.3060	977		979	. 557	196.	5.0	575	2.5	188.	.574		.3100	CAF	107										
7	Ì	? ï	-2.19	3.2		2.25	3.37	4.50				-	414	-4.50	2.1	-3.37	01-1-	\$	1.17				-4.51	•		•	6.20	*	8		*	8 9	-	-1.62	-2.15	2.7			***			3.6	2.79	1.24	2	5	20	2	
ž	;		. 392	2		.92	=	1.071	1.627	• • • •		01.	đ	•	•			.05	2	~ .	8	1.473	3	. 34	5	•	1.005	6	.53	. 151	.075	200				2		0 00.1	3				ž:						
5	•	7.50	3.135	1.372	-2.	902-4-	-6.649	-8.744	-13.273	7.306		7.700	ŧ	7.195	7.255		1	340	-2.429		-0.052	-1.725	7.332	200	5		7 1											- 2.340	5	4	8	-3.143	-2.30		599	197	2	1.023	
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43	į	•	**		2	2.	.679	199.	.673	. 64.		.4237	CAF	.740	.742	.79.	60,		-617		780	.764	. 14.	9048	3		000	. 513	25.	.521	. 523		515	. \$10	125	.516		.2500	*5	1967	340	.357		335	ž.	.353		.342	44
	•	£ . £ .	-2.20	6.7		7.24	3.36	***	*.*	24.42		*	ALP	-4.55	04.6-	-2.26	-	1:19	2.34	6 · ·	4 6	45.4-	•			;	\$.13 \$.13	0.4	60°E	1	<b>.</b> :	ī =		51-12	-2.19	-3.21	67.4-		41.	;		3.47	2.84	79*1	7.	.27	22	40.1	27.1.
ō		46.										1.20	3	920	- 69	400		.3	\$76.	60				2.00	3													4.50	3				ž						
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C.A.F.	;	.765	. 170			767	.766	101.	121	.75.		• • • 320	<b>C.b.</b>	#0Z.										• •3•00	CAF		*											.320C	CAF				.336						
316		4.4.4	-2.22	60.	1.15	2.20	3.40	<b>*</b> :	6.77	-4.47		~ *	***	9.4	£.	-2.2	50.	1.20	2.34	2.5	3.69				ALP	•0••	5.5	56.2	1.92	1.37	*	7	2.	1.7	42.2°	27.5-													
5		909		200	.293	.500	.721	.953			;	30	3	967										2.50 0	5	•	.770	1	-207		000	520-		. 25		į													
5		5.422			-2,380	-3.941	-5.758	-7.529	-10.982	50 70 70 70		9.40	5	7.873	5.024		*	-2.510			-9.10			4.930	5	4.631	***	-3.17	-2.073	-1-402	377	-10			2.380														
CAB	:		2	7.7	.237	. 241	.246	.254			į	r Z	CAB	.236	.232	262	.232	÷23	25.20	25.	250			z z	CAB	3	137		.135	* .	.133		761		.135														
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Table 16. Aerodynamic Coefficients for Configuration B1T4X3

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7.090	5	*60*	4.395	7.00				5.200	7.085	10.672	• 101	97.4	•	5		6.116	3.674	1.639	536	2.710	764.		***	*		4.930		5	7.180	2.01	4.67	9.4.0		-1-010	420	11.		1.972	2.552	20.															
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	46	4.3	-3.24	-2.18	3	: :		9	*	•••	7.7	•		ALP	1	-3.36	-2.22	1.0	•0•	91:1	2.32			.00				4.	8	Š	9.4	2.97		3	*	-15		2.1-	2-2-	22.7															
06.0*	CAF	.653	.659			2.5	104	***	.672	î	. 656	4320	2.00		.793		802	.727	.734	. 732	.720	101				3400		CAF	.485	**	į				ş.	24	2	.476		***	000	3	CAF	345	9	.342	980	245	326	334	.320		330	.323	
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. 4.520	3	051.0	4.092	3.674	1.102		-3.331	76.				7.700	5	. ;	7.302	3.307	515		423	-2.222	•	15.73		7.323		. 1.200	;	5	-9.672	-0.003		-3-230	-2.35	24.7			1.923	ž:		6.22	3.340	;	5	ž.	1	2	****	1	5	į:	.983		1.423		3.571
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Table 17. Aerodynamic Coefficients for Configuration BlT4X4

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Table 18. Aerodynamic Coefficients for Configuration B2T4X1

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Table 19. Aerodynamic Coefficients for Configuration B2T4X2

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Table 20. Aerodynamic Coefficients for Configuration B2T4X3

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Table 21. Aerodynamic Coefficients for Configuration BlT5X1

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Table 22. Aerodynamic Coefficients for Configuration BlT5X2

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Aerodynamic Coefficients for Configuration BlT5X3 Table 23.

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ž	243	.335	2	797	.272			ş	.335			:	5	796.	976		3	ş	*		914	900		;	Z	CAB	***	7		13		.135	.13	•		3	3	=	7.	;	ž	3	***	.05	500	ç	4	*	1		40.	90.0		,	
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	ALP	4.29	-3.22	-1-0	20.1	2.17	3.26			•	*	ı	4.4	•6.4-	-3.4	-2.10		21.1		30		P.92					ì	*:	::	60.6		 	Ę		79.	-1:13	-1-69	61.7	7			*	. !		*	2.01	5.		\$2.	2.		2.5	-2.29	-3-27	7
.3010	440	1	•			7.	.463	9	~ .	, es			4270		•		. 255		.594	600			.546	<b>\$</b>	į		9	. 3480	CAF	;		•	•	1		9	1	<b>!</b>	**		3100		4	.319	.322		716.	315	**	776	.00	.336	•	.324	.326
:	64.0	.327	<u> </u>	662.	**	\$ e ?	Š	.331	.365						S	424	9	.37	?		<u>.</u>	~	į	67.	£.			•	Ç	;		747	*	<b>!</b>	2		2.	25.	5 <b>4</b> 2		2		5	.107	2	•	6	-10	2:		2	6	ē:	102	. 103
\$.T00	5	;	3.236	ż		0.03	5.514	7.038		797			7.400		ž	3	1.239		264.		2	5.334	5.783	-8.594	*.634		•		5	;		2.0.2	.829	4	-1-767	502.4	-5.50	1	*		0.130		5	-4.228	*		***	783	**				1.333		. 10
0	3												. 0 . 0 . 1		Ś									1.000			•	. 0 04.1	3								017				8		3											170	
•	<b>*</b>	•													3									***					416		7:		6	į	-:				6		•	•	ž		*.7	2:		7	1	=:		7.1.	*	**	

Table 24. Aerodynamic Coefficients for Configuration BIT7X1

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190	IB CAF	200						3	•										3100										.352												
7.090 RM	5	205. 205								CM							102. 219		1.130 RM	CM CAB									200. 200. 200.												
	8	986 8.079					•			5							1.276 -9.512		•	3									269 1.582												
H - 1.00	4.	3.37							PC*1	ALP									и . 3.00	ALP									 												
9	CAF	466	9	7		**	įĘ	44		Š	70.	<b>? ?</b>	+14		.421	7	7	***	.3200	CAF	.375		976	5.5	 	ī:		.37		•											
- ##	<b>Q</b>	141.	135	132	261.	137	261.	•		3	-232	12.	7	12:	5.5	2:	22.	÷.	*	2									333												
4.750	5	5.652				-	-13-133										-10.01-		4.930	5	-4.804	-3.017	-2.101	-1.010	276	2	ě	1.23	2.403												
	Š		-	.325	166	1	.318		06.1								1.320		2.50 0	3																					
	41.6	4.4.	6.		2.7	3.			E	4	-4.52	40.0	70.7	ě	1.20	3.4	;	ş		ALP		.0.4	2.4	=	Ę	÷.	-1-17	3	-2.22	•											
.4010	42		77	•	9	ş	£5.		.4230	Ç	9	Š	? ?		45	2.5	27.	•	. 340	CAR	•	.41	ş		24	:	24.	÷	7.5	•	. 3200	242	7.		9		2	=	2	1	*:
=	5	5.5	Š	5 6	Ş	ŝ	įįį		:	5	46		2	777	.221	927	2.0	<b>*</b>	2	<b>87</b> 3	101	.192		<b>!</b> :	1	==	=	201	?		*	40	-02		0.00	620°	9	-02	620.		520
. 6.520	5	25.5	ž	-2.328	31	1	-12.022		1.700	5	3		11.	,	-2.662		-6.512	-2.649	5.850	5	4.97	-5-710	-2.973	-1.612	2	1	3	1.972	22.	***	2.490	5	-3.04	• • • • • • • • • • • • • • • • • • • •	-1:70			1.10	ţ		
•	đ	7	15	200	<u>.</u>	÷			1.10	5								.327	2.00	5									-		. 50			1	=	=		Ģ			8
	ALP	***		1:15	2.27					4			-2.25	= 4	:	2.33	3			414		2:5	8	*					97.6-	4.32		3		:		8:		?			
.3010	442	.321	331		980		Š.	• • 2 20	343	;	2.4	•	•		•	5	•	2	• 3600	S. C.	3	ş	6	~	? ?	***	: ÷	?;	. 9		.2900	840	**	ž	•339		124	÷:	3	9	
=	8	*61:	£ ?	-127		=======================================	7.	2		•	21	?	ž:		?	.243	8	Š	=	<b>C 4.0</b>	02.	8		4			É	**	?		:	55	*	į	Ş	į	90	\$ 6	,	46	ģģ
\$.700	5	****	2.603	.769	2	7.763	11.175	7.40	٠	5	7.80	1.653		1005	-6.709		-12-711	•	• • 200	5	-7-601	-5.67	-4-219	-1.0%		3.		::	3 9	;	2.340	5	-3.057		-1-736		432	===	į		
9	5						•	. 0 50.1									1.579		75 0	3											8	5	2								
ě								-	:										:												:	3	20.6								

Table 25. Aerodynamic Coefficients for Configuration BlT7X2

# ...

M = 1.00. 0 = 7.090 AN = .4190	ALP CN CM CAB CAF	-4.60 -1.021 0.020 .160 .428 -3.44750 5.903 .190 .442	-195 1.572 .145	.317 -2,364 .163	.598 -4.571 .149 .671 -6.702 .154	1.164 -0.626 .165	.041241 .143	N = 1.50 0 = 8.560 RM = .3660	870 80	902. 144.5 451	575 4-106 202	101 .735 .200	.152 -1.157 .196	#02 + #86+ #09*	.776 -5-605 .212 .944 -6.731 .221	7.17 1.525 -0.336 .236 .436 .12 .157 -1.85 .199 .420	3.00 G = 4.130 RM = 3100	CN CAB		.659 -4.314 .052	.510 -3.350 .045 .144 -2.431 .041	.232 -1.544 .039	.094624 .037	.026174 .037 040 .266 .037	-106 .705 .037	760. 609.1 642.												
060+ - MM -67.4 - B 54 M	ALP CN CN CAB CAF	-4-54 -1-054 0-300 -130 -4-29 -130 -4-21 -13-24 -13	-190 1.526 .111	.332 -2.529	.605 -4.574 .111	1.204 -9.352 .134	901. 156 640.	0017 MR 00000 . D 0011 . M	5	-671 4.862 .209		202. 2001 422-	137 -1.033 .200	#02' 250'R- \$86'	.924 -6.726 .224	7.13 1.379 -10.042 .251 .413 .11 .133404 .100 .427	H 2.50 0 = 4.630 AN = .3200	877 W7 W7		.467 -5.860 .102	.542 -3.674 .088	- 2.14 - 2.14 .074	176 -1.70	oro. 000 000.	025 .172 .071	***************************************	-1,271,451,441,411											
<b>.</b>	ALP CN CN CAS CAS	-6-30662 6-657 -120 -140 -8-366-18 6-6-18 -120 -140 -2-27807 2-662 -118 -177	-175 1.300 .110	111. 644.1- 182.	121 - 50.3 - 50.3	100 5/101- 000-1	cii. ki co.	M = 1.10 @ = 7.700 RW = .4230	ALP CN CF CAB . CAF	255. 904.7 464	1734 5.764	231 1-892218	- 100 - 100	215 E4E	1.033 -7.040 .243	7.04 1.510 -11.410 .27% .474 .03 .012 .011 .204 .475	A 2.80 6 - 5.890 RM3460	970 CM CM CMB C78		**** *********************************	.463 -5.276 .127	191 - 22122 - 191	117 -046 -144	501. 012. 020-	-100 .705 .140	-250 1.052 .140	-5.32 -597 4.155 -139 -609	H - 4.56 G - 2.450 RN - 13200	ALP CN CN CAB CAF	50. 64.4- 86.	272 -3.740 -5.34	23.	130- 1981-	120. 471 450.				-100 SILE 106
6194 68 694.6 - 9 69 M	643	100 01: 00: 00:0 00:0 00:0 00:0 00:0 00	#	196-1-062		1.362 -10.001		B - 1.65 G - 7.466 AB4220	44.0 CM CM CA6 CA6	400 020 0.000 1.000 0.00	200 0000	400 1201 1501 1501 1501 1501 1501 1501 15	.266 -2.123 .264	.955 -4-210 .200 .010 -6-257 .210	1.070 -0.270 .234	500 - 135 - 500	8 - 1.75 6 - 6.740 883600	842 42 43	48. 4	200 201 112-6-110-1 21-6	181. 020.0- 070.	.335 -3-776 -162	221 - 10.000	200	991			n 6 6 . 2.340 Ab2360		.730 -4-632 -629	20° 50°° 10°°	50° CR.2- CX.	2	200	Re	80° C10° C51°-	-2.32 -4.83 14.84 4.85 4.30	FF 555 555

Table 26. Aerodynamic Coefficients for Configuration B1T7X3

?	S S	1		Ę:			1	474	1		Į.		.4140		3	.433	1	į	;			1	į	3	•		. 3200	45		372			3		2			3	9																
i E	3	Š.	ž	2					282	.331	Ē.		=		3	1967	235	.23	Ş.			ž	×.	į	ž		*	CAB	;	, ç	.05		.040		ş	.040		1	150																
	5	7.28"	2.0	3.035	1.035					13.029			807.8		5	***	190	2.300	010.	577	2		4.742	10.383	45		4.930	5	. :	-7.019 -5.821	1.50	-3.372	-1.543	3	2	70	3.5																		
8	5	440	3	-367	127	į				*			9,1	•	5		1										2.50 0	5																											
•	414	9	-1.29	-2.10	-1.07	Į.	6:1	• • • • • • • • • • • • • • • • • • • •		4.72	7	!	•		P.	•		-2.19	-1.07	Ş			45.4		ş		£		i	6.03		6.2	98-1	1	26.		-1-22		****	2.4															
8	CAR	***		*	į	24.	•	7	•		9	}	0214		Ç			23	*	į	2	į		9	į		.3400	CAF	į	.413	***	517	24	174.		17.	214.			~1 <b>~</b> .	.3200	CAF		28.		. 335	225		.325	79.	110.	.312	.310	.321	
*	<b>5</b>	:	517	Ž.	=	•	Ę	=	2						5		ě			202	.27	Ę		?	ž	•	=	5		90	.053	\$	9	-69		Ş	9	, ,	90	ģ	2	5		3	9			60	6		50.	19	ē	Ž.	
6.78	5			3.18	1.222		-2.535	1	~		13.00				5	1	6.319		246		-2.071	3.0		10.331			5.030	5	;	19	-5.345	20.5	-1-7	-1.08	Ļ	į	1.701	2.30	4.070	5.133	2.430	5	•	5		-2-126			-	į	526		\$	3.27	
į	3										•			2:1	5		- 760	7	157	050	.23	?			11.	•	2.60		5	9:	į	7	· ·	3	9		215			76	4.50	. 5	;								2				
•	*				10.1-	.00	1.13	7.2	*:	*	3.5	•			46	!		16.6.		4	•	2.20	?:		*		=	•	ì	7		2.90		•	2:		-1.20	-1.7	2.2.7.	*	=	•	ì	\$.70		2.75	2:	1001	7.	70-	-1-32	6-1-	20.00	4	
610	CAF		• • • • • • • • • • • • • • • • • • • •	,,,,		3	.435	74.	**	.422	2		1	R	CAF	i	£;	* 1	3	.407	.497	9	Ž.	3	2.		*	243	į	524.	524	*	*	• • 3 5	2		**	ž.		Ç.	.2500	CAF		516.	.313	910	į.	Ż.	202	£	ē,		5	ķ	
:	642	!	1:	2			1112	.113	.110	.127	3	131		•	3		. 22	Ę	242	2	.202	Ş.	į		325		:	5	į	•	2	5	į	560			7	\$	įį	ė	=	3		31	3	9	Ş	.043	7	3	ī	7 3	17	į	
4.520	8		. 323											7.780	5	,	6.97		1.522	***	2.591	4.34	4-127		***		1.700		;	207.0		***		1.97	2:		1-702	2.530		6.43	2.340	5	•	100	-3-174	*	200-1-	\$	ě		•		84.2	3.40	
•	5		6								•			0 0:.	5										1000		. 2.		5	•									?			5		.792		380	**	8	8		5	2:	•	527	
	*	Ì	4	-3-20		9	3	2.23	3.33	•••	6.65	4.37		•	•	Ì	7	-3.32	7.7.7		1	2.27	3.34	3	1				ì	• 1			60° Z		ŧ	\$ 3	- 1	-1.61	77.5	1	•	•	ì	*		2.02	=:		2:		-1.20		27.5	-4-27	
0100		•	.343	. 365	.365	2.	7.				.337	.350			.4270	945	į		7	23			016.	Ž.	ŝ				.3040	ż	į		į	į	24.	Ę.		3	\$		. 3100	545	į	46.	398	.351		200	186.	35	980	3.5	98.	.353	
			•••								***	•			:	3	Ì	.33	-	Ę		Ę	ě.	0.	2:		•		=	3	:	1	=	2:	*	-13	1		*			•	į	160		.00	3	9	Š	1 9	ě	3	;	į	
4.700		5										•			7.40	:	•	7.094	5.070	2:				004.4	2	7.983			. 1.340	5			3.972	•	7.007	-4.095	Ž.		25		¥1.4	:	5	4.013		2.001	-1.77	747			ž.	1.70	r.	4.333	
9	• ;	5													1.00	ŧ													• 05.1	3	•		*			Ĵ	2				00.6	٠	5	.050	ê	*	į		~	9		5.	~	•	
•		,	16.4-	- 3.22	-7.7	-1.05	ō.	7:15	2.7			7			:	;	Ì	-4.42	-3.31		3		7.7	3.30						**	Ì		~	-1-10	ŧ:		3.3	?	į				ì		;	2.03	2		~	~	-	-1.01	26.5	1	

Table 27. Aerodynamic Coefficients for Configuration B1T7X4

.4140	CAF	.537	. 556	255		. 36.	3	745		0914.	CAF	-512	. 521	.519	• 524		808	200	. 530		.3100	3	976.	.375	.352	346		,		.352			9	. 353	•
: ?	<b>8</b>	.536	;;	.539		. 532	65	3		2	842	.182	25	:	51.	12.	.189		. 150		ž	6 <b>V</b> 9	50.00	200	-01	50.	2 3	3	3	150.		.055	.050.	2057	ł
7.090	5	5.630	3.897	*82°-	4.097	4.277	4.539	7.729		. 400	5	****	5.073		570	2.393	÷:1:+	1,041	452		4.130	5	-8.624 -7.478	9.326	.5.190	4.074	2.00		~	401	2.5	1.173	1.640	3-163	
2		659				-		•		0		012				•		•			•		1.237	•	•	•	•	٠.							
. 1-00	_	55.5								H = 1.30		. 64.43					_				H . 3.00	_	***	_	_	_				_					
		•	٠,					•				1	•																			•	•	•	,
٥	ų	~ ₽	F 8	~	•		•	0 1	,	c	u	•	<b>.</b>		•	. •	•	۰~			•		•		_		•				~.	•			
3	3	24.		•			*	7	•		<b>4</b> 7	Ÿ;		ž	Ý.	į			ż		. 3500	3	9				Ę	7			200	ì	ě.	Ċ.	•
*	3	3.0	7			Ŷ	. 423		•	<i>3</i>	5	97.		*	÷.	7.7.	. 253		.265		2	5	\$ 5	6	-0	9	.073	\$ :	1	.049	790	200	290	Š	9
¥.14	5	7.004	3.971	597		.73	*			•110	5	4.074		. 50	2	200	5.798	1.320	7.0.		5.5	5	129.0		5.023	1	3.43	2.32		.53	•	717	2.5	3.503	
	_	100			_		Ϊ.			2	3							•			9		1.340								,				
•		***	•							1.20											\$:2	3	3:				_				_				_
•		77	77			-	•	• ;	ĺ	=		7	77	7	•	~ ~	•	••	7		×		••	• •		•	•	~	-	•	٠	٠ ٦	71	77	1
3	4	£1.7.	664	ī		•				0£24·	3	360	3	2	. 572	55	3	576	i		.3500	3	8		?	-		9	•	3	ě	į	ř.	į	346
:	5	757		*	•	.217	?	0		2	3		3		ŝ	į	?	9	į÷		į	3	121		=	5	=	•	į	į	Ę		9	9	=
0.4.0	5	4.342	3 6	610		3	285	÷		7.700	5	***	£		î	į.		22.	254		• 200	5	2		ž	2	249	= 1		2	25	ž	=	â	•
æ	_							١.		•	_		_								•		00 -12.996												
•		26					_			1.10		-	;	77		7	•	•			:		004-				•	•							
i	4	56.4.	-9	0	⊸.	• ~							•	-	. ~				• •			•						•					3.2		į
		77	~ 7	•	٠.	: .	;				4	7	-3.27	~	6	3:		*	7			1	-		.5.5		3.37	÷:		•	Ġ.	-	77	•	•
		77	77	•		: ~	;			•	4	7	-3.27	-2-17	6	2:	***					**			25.5	4.44	3.37	£		•	Ġ.	- 7	7	**	•
9.10	447	300								• • 0220-	CAF	•	•	•					200	2.6.	. N 0466.		100									• •		•	•
	CA8 CAP	.340	.350	.357	121	.376	.371	. 129	• **	*** .4220	CAB CAF ALU			106.	2.	*		980	·		AN3840 H .		•		524	***	*			**	198.	0.7-	1 (	•	•
•	445	.306 .337	.266 .359	.295 . 357	200 - 1000 200 - 1000	269 .376	.204 .371	100 E 100 E	*** 11#.	- ** - ** - *** - ***	Cm Cas caf at			196. 98.	***	***		986		į	8.560 4h3860 H .				122 .475	****	***	****	117	.191	106. 711.	0:1-		• •	•
0146. ************************************	445	5.814 .304 .337 4.390 .288 .340	2.501 .200 .350	.072 .295 .397		45.049 .376	-9.000	-7-246 .313 .337	14. 60. 18. 18. 18. 18. 18. 18. 18. 18. 18. 18	4 7.460 4% .4220	CH CH CAB CAF ALL	. 575. 500. 618.0		2,000 .000 .000		-2.077 .400 .594	986. 984. 084.61	-7.877 .668 .588	000 000 USB 111		6 = 8.560 AB3860 B -	CH CAB CAF	7.042 .129 .461	FUA. CUT. 100.	1.926 .122 .475	-150 -120	-2-397 .116 .494	100° 521° 658° 9	11. 12. 14.0	-11.073 .191 .904	-2.400 .117 .901	9.7		•	•
. 5.700 AM .	445	.306 .337	2.501 .200 .350	.072 .295 .397		45.049 .376	-9.000	-7-246 -313 -337	14. 60. 18. 18. 18. 18. 18. 18. 18. 18. 18. 18	•	70 CAS CAF 42.	. 172 6.973 .005	369 6.032 579	. 133 2,500 . 454 . 551		247 -2.077 -460 -594	986. 484. 088.4- 044.	.023 -7-877 -000 -500	000 000 100 11 000 11		٠	CN CN CAB CAF	023 7.042 .129 .403	### 100 000 000 0000 0000 0000 0000 000	120 1.12	**** 02155- 980.	280 -2-307 - 116 . 486.	104. 421. 000.4- 216.	151 - 151 -	1,425 -11,473 .151 .504	106. 711. 000-5- 605.	9:4-			•
4 5.700 AM .	445	5.814 .304 .337 4.390 .288 .340	-, 306 2, 501 . 264 . 359	855. 878. 611	100° 102° 000° 000°	02. 600.00 . 200.00.00.00.00.00.00.00.00.00.00.00.00.	.061 -5.045 .204 .371	100 000 000 000 000 000 000 000 000 000	116. 809.8 - 100 700	•	שרם כש כש כשם כשם שרו	. 172 6.973 .005	369 6.032 579	. 133 2,500 . 454 . 551		247 -2.077 -460 -594	986. 484. 088.4- 044.	.023 -7-877 -000 -500	000 000 USB 111		•	CH CAB CAF	7.042 .129 .461	### 100 000 000 0000 0000 0000 0000 000	120 1.12	**** 02155- 980.	280 -2-307 - 116 . 486.	104. 421. 000.4- 216.	151 - 151 -	1,425 -11,473 .151 .504	106. 711. 000-5- 605.	9:			•

Table 28. Aerodynamic Coefficients for Configuration BlT8X1

	3	\$	1						ķ	š	•			•	ŧ	•																						
z	CAB	*:			:			• • • • • • • • • • • • • • • • • • • •	•11•	•11•	110			2	2	0		=																				
4.130	5	-5.533						2	- 203	-264					2.130		1000	100.4																				
3.00	3	- 750-	•			•	•																															
	¥	5.43					67.1	?	2.					-1.63	-2.34	*		14.38																				
.3200	CAF	. \$29		188	.573	*	. 56.	. 572	7	7		***	. 55	į	3	3	. 22	ž																				
Z	4	.156	ž	.153	•		=	171				:	-	174		7	-	\$																				
. 4.930	5	-5.634	-4.020	-3.702	-2.170	-1.010	-1.273	807	1				Į	3		2.013		6.010																				
2.50	5		.733	£5.	2	.273	.192	133	ŝ			į	- 165	1		-	451	119-4																				
	ALP	80.9	5.6	4.03	7.00	1.0	1.39			:	-	-	-1-10			-2.2	-3.24																					
	5	1	. 537	3		ž	47.			•	į	į	1		•	3	. 460		į		.3200	CAF	3	•											2	7		ì
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Table 29. Aerodynamic Coefficients for Configuration BlT8X2

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Table 30. Aerodynamic Coefficients for Configuration BlT8X3

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8.8			×	į	Ų.		•	•	7	= 7		•	•																
	¥	8.5	*	2.63	-		*	2.	2	7:			*																
. 3200	3	.553		ĭ	3	<u>.</u>	2 3	ł	2	3	25.	2 2	285	<b>:</b>															
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8 · ·	5	-7.174	7		-2.780	: :			260	**	2.019																		
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		8.			•	1.33	ą			7	‡.	**		Ŗ															
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Table 31. Aerodynamic Coefficients for Configuration BlT8SX1

.3130	CAF	1	77.					430	714.			7	7.7.7	77.7		. 432	.63%			614																												
z v	840		775	771	771.	. 17.	.127	129	1 20		?	2	7		1611	131	=	101	.131	7	:																											
. 4.130	Z.		-6.607	-5.451	167.	-3.122	-1.975	-1.384		700	***	.245	. 20		1.340	3.905	7.7	5,4.7	3.531	484																												
9	č	;	.984	0	.628	094	.290	204		171.	ó	036		2	-197	2.281		365	522		043																											
9.00	0	ì	96.9	16.4	3.93	2.88	1.84	5	2	?	97.	7.			1.27			2.32	3.17		4.33																											
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.3200		3	694	4.	.47	4		•	7.	4	•		•	•		3	₹.	1		•	4	•																										
ž		2	. 157	4.3				•134	.127	-124		.163	.123	.122			.124	40		. 129	130																											
0.930		5				,	0000	-2.326	-1.457	0.0			•	780		:	2.045		6000	3.634																												
9	1	₹	;																																													
2,50		A.C.	:		5		.01	56.1		:	•	.36	41.0		•	1.19	,	7.1	5.25	1.75		?:																										
•	•		•													•		•	•	•		•																										
946	?	CAF		. 585			.582	. 592				. 593	904		9	200	4	• > 40	3	4		ć	•		.3200		44)	-	.372	. 356	. 361			. 308	.361	151			. 30	.353	.353	350	Ş			.35		
:		640		.175	. 169	.161	. 5	157			-	*			97.	. 147			94		. 12	75			ž		CAB	,	.040	6	.047		,	•	940				•	440.	*	990	940			• 0 • 5		
		5		8.265	6.689	5.454	4.023	2.400			25.			5	164.	000		2.280	7.981		. 343	4.717	;		2.450		5		-5.575	7.	2.416			. 545	1.042				2.	099	1.042	3			2.90	3.88		
	•	Ž		1.156 -	•	•	•			•	•							_			_				•		Š		. 654.			١	_	•					_	_		_			_	_		
	2.00	4.4		1.17 1.							_										_				H . 4.50		4.6											_	•		•				_	•		
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		340		i	.636	000	,			. 655					•			. 643		.073	.692		. 00		2,000		447	•	306		.36	.379	376			.33	.378	378	*			. 343	.367	.35	173			
	e Z	5	;	191	.175	0.1.0			:	.162	. 41			. 163			•			•	4		•				5	į	740		•	1047	140		9	-	190	400				-00	.00.	.067			.00.	
	8			4.825	765	5		,		.078			î	167			3		,	2					5	•	;	,		5	•••	824			į	.162	3	. 02		. 233	*21.	. 176	619	0.76		***	.00	
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Table 32. Aerodynamic Coefficients for Configuration BlT8SX2

.3100	S.	.436	.431	.431	435	0.4			. 434	. 429	6430	72.7			•	.437	(44																						
	CAB	106	106	105	104	400		60	103	103	103		2	103	103	104	301																						
4.130	5	5.665																																					
3.00	Š	. 972	•						_	_	_			_		٠.	_																						
	ALP	5.95	3.80	2.85			7	:	÷	25			97.1-	-1.82	-2.32	-3,32	35 42																						
.3200	CAF	. 489	2		.403	.495	164.	. 697	5			-	.485	.487		7.		.487																					
2	3	.165	•	5	2:	. 169	. 169	4	9			. 169	. 169	170			2	.171																					
- 4.930	*	-8.007	-6.641	-5.213	-3.0%	-2.478	-1.738	100	16.4		. 551	•••	1.541	200			3	5.520																					
2.50	3	1.001	668.	.702	.517	.330	.733	*			054	-119	208	206			-, 20	749	•																				
	41.6	10.0	\$.0 <del>\$</del>	90.4	96.7	1.92	.13		•		-:-	2.	-1.21		*	-2.2	-3.5	-4.2																					
.3400	CAF	. 555		Ŷ.	00.	. 586	.502	. 582	.547		700	. 202	. 592	.586	9			. 572		. 3200		CAF	.371	.363	2	.365	356	356						.306	.340	*	305	.373	
z Z	CA8	. 208	. 20 I	.195	:		64.	641	9			•	3	181				.182		Z		<b>6</b>	•045	.040	.045	**0*	***	440							0	0		**	
5.850	7,	-9.324	-7.67	-6.172	-4.547	-2.965	-2,087	+1.796				0.00	1.807	> 404				6.526		2.450		ŧ.	-5.769	-4.672	-3.634	-2.630	-1.691	10.						1.00	1.533	2.028	2.98 I	3.0	
9	3	1.217	466.	.798	.584	.301	.768	147	4		028	-135	233	46.				848		9.50		3		+69.	•539	392	240			901	5	*20-	260	139	226	29	-: 445	590	
	41.9	6.12	5.08	\$0.4	5.99	*	1.30		:		-:		-1.21	-			- 3.4		•	ř		ALP	5.92	4.92	3.91	06.4				:		=	:	-1.21	-1.72	-2.22	-3.22	-4.23	
			. 423	678	7			92 \	656	672			2	100	299	657	•		669	00.6	,	5.45	;			9,0	706	371	369	.357	336	. 396	365	.393	375	105	362	.369	
,	•		2											_		_		_	_	4		CAB																0.09	
						•		•												9		*			•		-												
								-2.42	-1.57	71	-	•	1.00	1.87	2.76	44.4				•	,				•	•	•	•	٠									* 120	
	• (	5 9			1			. 301	967.	600			121	237	-, 150			7	693	· ·		5	1				•	.282	.173	. 102	.039	037	- 105	174	4		4	010	
		3					•		•	•			~ •	-1.13	99-1-				62.4-	•		41.9	;	600	•		7.61		1.20		\$77		11			9		0	

Table 33. Aerodynamic Coefficients for Configuration B1T8SX3

	CAB	101	. 60	500		063	200	0	2			Š.	.093	400		•																					
4.130	5	-7.030	•																																		
3.00 0	5	7. 986.				_				_	_	_	_			_																					
1 · 3.	414	5.45	3.92		7.00	1 2 4		7.	77.	•	-1.26	-1.78	-2.27	7.7	-3.6	-4.3																					
.3200	CAF	184.	6.		110	700	! 8	2	. 465	•	• • • •	. 488			•	**																					
ž	CAS	131	.127	.12:		-71.	:	*71.	.124	.12	*71.	**			•12•	.126																					
0.6.4 = 0	5	-6.283	-5.306	-3.900	-2.30	-1-1	-1:101	*	.218	Ē	1.541		117.7	7.1	: :	5.50	:																				
2.50 0	Š	1.001	.687	š	.323	.230	- 142	8	028	116	200			372		11.73																					
	ALP	4.04	5.	56.2	•	1.38	•	.3	•	4	-		7	-2.22	-3.23	4 23																					
.3400	CAF	.598		ĝ.		. 195	3	3	ŝ	9	705			8	. 5 98	9			.3200	CA	;					.332		.353		100	.395	0	20	.363	.358	.364	
ž	CAB	204	ź	. 205	.204	•204	2	202	202.	202	2	•	?	ě.	ê	200	•		=	5	;	8	2	2	8	0	\$	ò	0	Q.	80.	0	•	ģ	\$	•	
5.850	5	966-6-	6.33		-3.022	-2.139	-1.342	522	. 125	1.024				3.457	5.015	4.444			. 2.490	5	:			- 2	-2.677	****	-1.145		227	. 225	8	1.182	**	2.117	3.073	4.074	
2.00 0	5	1.1	20.	. 575	.371	. 263	:	100	027	4717				427	623				0 05.4	3		î		•26.	~	<b>%</b>	.163		.032	031	90:		*?	303	445	586	
	416	•	***	2.97		•6:1	į	.34	15				:	-2.23	-3.25				•	474				9.63	7.81	9.	1.25	.75	*:	**	*	-1.28	-1:3	-2.27	-3.28	-4.7	
900	*	:		9	.00	~		4	1			•	?	1		į	•		.2500	<b>1</b>		.346	.377	.377	.374		.371	.352	.361	.340	.351	.356	345	.354	196		
:	3			707	* 50.	-200	- 288	. 28.7				2	£ .	784			. 2		:	5		*00	•	•	10.	***	***	•	•	••0	•	**0	100	900	•	4	;
200		10.90	. 5	5.57	3.752	2.770	1.053		120		0	* 1.	2.643	1.667			7.111		2.340	5		190.4	\$	3.852	2.805	1.003	1.232		276	•62.	.727	1.204	1.701	7.197	3		•
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Table 34. Aerodynamic Coefficients for Configuration BIT10X1

.3100	Z.	444						9 5	9	ŝ	555	3	8	262	252	551	. 565																
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2	CAB	Š	č	č	d	•	5 8		•	ě	90.	ě.	ě.	8	80	0	0.																
4.130	5	A. A. A.	1	. 70.							.314	1.085	1.028	2.570	3.342	4.770	6.322																
•	₹	717	.023						707	200	.043	.150	.253	356			:																
3.00	ALP																																
	•	4				-	: -	:	•	•	•	•	7	7	-2-	.3	4.4																
.3200	CAF	79	4	90	9	1	3		8	ŝ	.603	609	409	3	9	200	. 593																
z z	<b>CA8</b>	141	130	136		5			128	.129	.129	•129	.129	.129	961	.133	136																
4.930	5	2	200	ŝ	9				2	9	5	001	111	•	137	603	2																
•		•	•	•	•	•			•								6.701																
2.50	5	14.35	11.11	.87	1	-			= :		-03	-1.	2.	- 36	94.	70	932																
	Ì	*	5.11	40.4	2.00					•	•	5	-1.22	-1.72	-2.27	-3.28	4.33																
.3400	CAF	.704	*69*	٠ 8	. 692	689	•	789-	. 663	***	ì	700			•	•	.683	.3200	CAF	***			***		4.	•	84.	.473	.472	*4.	.465	.472	
ž	CAB	.221	.21	.221	.224	. 221	.223	. 223	. 223	333	,	,	77	***	- 223	977	*22.	z	CAB	910		9	9.0	0.0	0.03	000	3	140.	.04	10.	.04	040	
5.850	5	10.537	-8.657	-6.78	-4.964	-3.209	-2.263	-1.44	583	24.2							1.200	2.450	5	740	5.50		104	-1.967	-1.309		301	.278		1.414	1.946	2.429	
2.00	5	1.432	1.173	•	• 13	.433	.307	.196	.079	-1035	*							0 05.4	5	500		9.4		. 287	. 200	.125	.042	038	124	- 20 t	283	357	
~	ALP	4.22	2.15	•	0.0	. 6.	·•	÷	.3	41	7		***						A.	40.5		3-15	2.07		1.26	.17	•25	3	:	-1.27		-2.27	
.3600	442	•1.	. 733	. 73	. 737	. 739	.73	7.	142					?	?	?	. 719	.2500	4	. 510	84.	۶.	*	.516	617.	.493	27.	£ .	11.		~		
2	3	•	•	:	•	•	.103	. 103	.182				•		2	707	~	ž	C A 8	.051	.051	ş	150.	.051	.05	160.	250	200	200		200	760	
• - 200	ŧ	11.903				ì	004	•	711	243	217				1	3 3	.20	2.340	5	•	1	547	277	••0	465	+0+		9	9		2.00°	•	
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Table 35. Aerodynamic Coefficients for Configuration BlT10X2

.3100	¥.	555	* * *		100	220	550	555	663		000	240	700	44		6.0	220	246																								
•																																										
ž o	5	5 90.	5.6		5	ě	80.	80.		5		6	0,		5	•		Ö																								
4.130	5	-9.36	-7.768		-4.537	-2.945	-2.073	240			336	1.128	1.078			3.547	5.059	A.708																								
3.00	3	1.250	1.027	21	.595	.384	176			800	**0.	147	264		307	466	668	- RA7																								
	ALP	0.00	4.97	3.95	2.89	7.67	44		70.	• 5 8	23	72		7.	-1-1	-2.28	-3.23		7																							
.3200	¥.	.593	. 597	3	507	ç	3	8	3	.610	200		710.	3	- 602	101		3	8																							
ž	CAB	120	. 117	1111	101			• 103	701	101			-102	.03	104			-	907.																							
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Table 36. Aerodynamic Coefficients for Configuration BIT10X3

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5.450 RN .		. 137	131	• 129	.122	122					2:	.133	::		• 120	.128	0817	:	161.				960	2	8	050	\$	640	\$0.	640	\$	9					940
G . 5.850 RM .	<b>9</b>	-13.594 .137	-11.302 .131	-8.924 .125	-6.517 -122	-4.234 -122	121	****	-1.653		27.	1 444	100	27.1.2	3,945 .126	5.103 .128	7.274 .130	181	4.675 .131		0 = 2.450 RM =	CAB	050- 050	900	050	474 .050	-2.240 .049	-1.571	640	262 .049	240	440		600		2.695 .049	1040
2.00 G . 5.850 RN .	CN CH CAB	1.408 -13.594 .137	1.328 -11.302 .131	1.046 -8.924 .125	744 -4.517 .122	46.234 122	121	****	.216 -1.033	ATT - 184 - 640*	201. 954. 150	122	75. (50.) 76.	-324 2.1.1	921. 3,945 .120	597 5.103 .128	130	101 101 101 101	-1-137 4-625 -151		4.50 0 = 2.450 RM =	CM CAB	. 050 - 7, 408	050- 911-9- 050-	050	050 414.4	200 -2,240 -049	900 -1-571	128 871	040 - 345	940 040	970		261-	278 2.077 .049	360 2.695 .049	940
G . 5.850 RM .	CH CAB	1.408 -13.594 .137	-11.302 .131	1.046 -8.924 .125	744 -4.517 .122	46.234 122	121	****	.216 -1.033	ATT - 184 - 640*	201. 954. 150	122	75. (50.) 76.	-324 2.1.1	921. 3,945 .120	597 5.103 .128	130	101 101 101 101	-1-137 4-625 -151		0 = 2.450 RM =	P CN CN CAB	. 050 - 7, 408	050- 911-9- 050-	050	050 414.4	200 -2,240 -049	900 -1-571	128 871	040 - 345	940 040	970		261-	2.077 .049	360 2.695 .049	940
N = 2.00 G = 5.850 RN =	CN CH CAB	4.04 14608 -13,594 -137	4.01 1.328 -11.302 .131	1.98 1.046 -6.924 .125	3 43 .744 -4.537 .122	122	16. 676 c 777	1710 (p. 67) april 160"	FIT (60.1- 912. +0.	ATT: 194°- 640° 06°	2720053 .454 .120	123	760 760 7610 7610	*71. */1.5 *5E. *5.1-	021. 3,945 194 77.1.	12597 5.103 .128	130	101 0 100 0	-4.34 -1.137 4.625 .131		H = 4.50 G = 2.450 RM =	P CN CN CAB	. 050 - 2.508 .050	060* 411.4- 554.	0500 998 90 800 1	0500 445.51 950	640 042 2 40 1	640" [25] - 906	640 [24" 95" 10"	900 200 - 000	600 092 980 0	670 ere	210 011- 21-	270 /6401 261- 271-	278 2.077 .049	-2.23360 2.695 .049	970
.3600 H = 2.00 G = 5.850 RM =	CAF ALP CN CM CAB	4.04 1.04 1.04 1.13,594	247 4.01 1.328 -11.302 -131	125 - 6.924 - 125	3 43 174 4.537 1.122	27. 46. 48. 48. 41.2	TARE STATE TO THE TARE TO THE TARE TO THE TARE T	144 CACA BEEN BEEN BEEN BEEN	11. ECU.I- 012. +0.	411. Ind. 970. OE. E47.	745 -120053 -454 -120	200	100 (100 TATES 61 - Q6)	-730 -11-20320310	.737 -1.40 3.945 .1.20	727 - 128 - 540 - 540 - 128	ORL TAR OF THE CONTRACT OF THE	THE PROPERTY OF THE PROPERTY O	151. 550.9 781.1- 46.4-		* .2500 M * 4.50 G # 2.450 RM #	CAP CN CH CAB	0.50	050 411.4- 454. 44.	050	050 458.81 854 44 4	0+0° 040° - 100°	640 151 906 151	640 [24" 46" 46"	940. 040. 00.	6+0° 096° 460° 06° 544°	67C 6F4 750 17 17 17 17 17 17 17 17 17 17 17 17 17	20° 21° 21° 21°	237 -1.22 -1.52	-1.73278 2.077 .049	-2.23340 2.695049	970
NN - 3960 NN - 2,00 G = 5,850 NN -	CAS CAF ALP CN CN CAS	751 408 -13,594 4137	TRUE CONCILL SCALE TO SELECT TO SELE	821 - 426 61 440 1 80 M	200 122		TATE OF THE PARTY CALL TO THE	131 AAA37 BAGO EGOT EAL BATO BATO	PART - NEW-11 - 012	*14.3 .30 .07.9 .34.3 .34.3 .34.3 .34.3	140 745 -20 -053 -456 -120	201 979 1 000 11	100 - 100 - 1010 - 1010 - 1000 - 1010	-146 -739 -1-24326 2-159	127 .737 -1-77461 3-945 -120	140 717 -2.40547 5-103 -128	ORIGINATION OF STREET	THE THE DESCRIPTIONS OF SERVICE	161. ECO.V 781-1- 46.4- 767. 001.		AN # 2500 M # 4.50 O # 2.450 RM #	CAB CAP ALP CN CH CAB	0.00 20 100 100 100 100 100 100 100 100 10	060* 911.4* 558. 44.		CHCC WAS SERVICE AND A COLOR OF A	040° 042'C- 86'C	0400 EAST 0000 FE T	6404 (544) 461	940° Z46" #40° #C	600 000 000 00	44C 44C 46C 46C 46C	100 Page 101 - 22-1	.007	**** 7.0°5 875 1.1°5 276 2.077	.067 .47523340 2.495 .049	870
# 4.200 RM # .3400 M # 2.00 G # 5.850 RM #	CAS CAS ALP . CN CM CAS	751 408 -13,594 4137	TRUE CONCILL SCALE TO SELECT TO SELE	821 - 426 61 440 1 80 M	200 122		TATE OF THE PARTY CALL TO THE	131 AAA37 BAGO EGOT EAL BATO BATO	PART - NEW-11 - 012	*14.3 .30 .07.9 .34.3 .34.3 .34.3 .34.3	140 745 -20 -053 -456 -120	201 979 1 000 11	-142 - 140 - 141 -	-146 -739 -1-24326 2-159	127 .737 -1-77461 3-945 -120	140 717 -2.40547 5-103 -128	ORIGINATION OF STREET	THE THE DESCRIPTIONS OF SERVICE	151. 550.9 781.1- 46.4-		* .2500 M * 4.50 G # 2.450 RM #	CAP CN CH CAB	050. 608.7- 050 : 20.4   105. 240. 444.7.	000 411 41 600 600 000 000 000 000 000 000 000 00	ONO STATE OF THE PROPERTY OF T	CONC. AND AND AND AND AND AND AND AND AND AND	10-10-10-10-10-10-10-10-10-10-10-10-10-1	040° [15] 000° C C C C C C C C C C C C C C C C C	OPPORT TRANSPORT OF THE PROPERTY AND THE	000 COMPT 000 00 000 0000 0000 000000000000000	THE COLUMN TWO IS NOT THE COLUMN TWO IS NOT	640 - 640 -	200 Pier 210 Per 200 P	1.564 .067 .532 -1.22 -192 1.437 .017	2-258 .007 .482 -1-73278 2-077 -044	2,985 .087 .475 .2.23360 2.695 .089	
6-200 AN - 3960 N - 2.00 G - 5.850 AN -	CA CAS CAS ALP CN CM CAS	A.04 13,594 - 13,594 - 13,594 - 13,594	1810 CON-011 070 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	SOUTH CONTRACT STATE CONTRACT	-10-509 - 1-10 -	100 100 100 100 100 100 100 100 100 100	101 - 101 -	111 APROVE BARO MAIN MALO MALO MALON	Part 194 - 744 - 7	11.230 .140 .743 .30 .079 -101 .11.7	745 140 745 -20 -053 -454 -120	200	100 CADA 2010 DIST BIST DEL COLUMN 2410 BEC-1	2.631 .146 .7391.24324 Z.174 .124	120 246.5 1040 - 71.1 - 120 1 3.945 - 120	2 - 2 - 3 - 12 - 12 - 12 - 12 - 12 - 12	ORDER SECTION AND CONTRACTOR OF THE PARTY OF	100 1 100 100 100 100 100 100 100 100 1	161. ECO.V 781-1- 46.4- 767. 001.		AN # 2500 M # 4.50 O # 2.450 RM #	CAB CAP ALP CN CH CAB	050. 608.7- 050 : 20.4   105. 240. 444.7.	000 411 41 600 600 000 000 000 000 000 000 000 00	ONO STATE OF THE PROPERTY OF T	CONC. AND AND AND AND AND AND AND AND AND AND	10-10-10-10-10-10-10-10-10-10-10-10-10-1	040° [15] 000° C C C C C C C C C C C C C C C C C	OPPORT TRANSPORT OF THE PROPERTY AND THE	000 COMPT 000 00 000 0000 0000 000000000000000	THE COLUMN TWO IS NOT THE COLUMN TWO IS NOT	640 - 640 -	200 Pier 210 Per 100 P	1.564 .067 .532 -1.22 -192 1.437 .017	**** 7.0°5 875 1.1°5 276 2.077	2,985 .087 .475 .2.23360 2.695 .089	870

Table 37. Aerodynamic Coefficients for Configuration BlT111X1

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Aerodynamic Coefficients for Configuration BlT11X2 Table 38.

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8.8			_	_			_					_																	
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.3460	CAF	. 555	ij	. 365		505	.571	3		926	. 557	. 555		.3200	CAR				.422	417	.424	.423	.42			435	***	•2•	
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Table 39. Aerodynamic Coefficients for Configuration BIT11X3

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Table 40. Aerodynamic Coefficients for Configuration BlT12X1

H = 1.00 G = 7.090 RN = .4190	CN CN CAB	-4.29273 1.405 .171 .547 -2.21186 .916 .169 .549 -2.13106 .443 .169 .549 -1.05036 .150 .167 .557	0022065 .165 069180 .166	7 .227 -1.006 .169 7 .324 -1.561 .167 5 .514 -2.626 .181	M = 1.50 G = 8.560 RM = .3860	CN CM CAB	337 1.710 .150	246 1-250 -150 158 -806 -145	074 .401 .143	141, 660, 400,	.169752 .150	.262 -1.225 .161	4.49 .362 -1.743 .168 .436 6.78 .590 -2.952 .187 .425 .02 .007 .016 .142 .455	M = 3.00 G = 4.130 RM = .3100	ALP CN CM CAB CAF	.790 -3.784	.640 -2.975 .111	011. 646.5- 616.	1313 -1.364 108	-228 -1.002 .106	.149652 .105	.070412 .104	.031141 -102	101. 610. +00	042 .107 .101	062 .365 .101	158 .691 .101	707	
RN4090	_	.108 .510 .105 .521 .102 .526			AN = .4160	_			_				.197 .501 .156 .536	4N = .3100	. CAS CAF	.140 .273					_			_		_	_	.122 .272	_
05.0 = 0 .750	5	-4.29285 1.492 -3.20186 .894 -2.12095 .371 -1.05042 .202	200	22.5	. 1.30 0 - 6.400		7.24	100	96	5	102	200	73 .554 -2.730 05 .036 -171	2.50 0 0 4.930	_	\$11.	-637			.238	.150	•		8	032			_	- 31
0107		.487 .485 .495 .904			£ 230								000	. 3500		.310												-321	•
. 4.520 44 -	C# C48	1.362 .101 .765 .099 .270 .096			. 7.700 RM -	643 N3		•					-2.407 .214	. 5.050 AN	0W CA8	-3.003 .176				_						_		1.000	
\$ •	_	-4.27270 -3.19172 -2.11003	<b>-</b> -		01*1	_	_						060.	2.00	***	1.00	_			_	_			_				-3.15227	
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Table 41. Aerodynamic Coefficients for Configuration BlTl2X2

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Table 42. Aerodynamic Coefficients for Configuration BlT12X5

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## 0550 ##	842 P.	195	.6%	200	31. 56	-1-122 .105	-1.500	-2.122	. 181	9 - 7.700 RM -		.303	1.073	į		317	000	1	-2.680		**** 05.0 0 . 0 . 00.0		-4.416	-3.710	-3.0%	-2.49	-1.967	-1-404 · 104 · 1-	111	1011 1161	202	040	.187 .172	.410	3		
0 - 6.520 AM -	870 HJ NJ 4	1,000 ,195	et	.048 -382 .34	140 996	190 -1-122 .105	.273 -1.996 .195	221.22		H - 1.10 G - 7.700 RM	5	1.568 .303	199 1.073	121	****	-076 317	.141	290 -1-447	.502 -2.680	670.	M - 2.00 Q - 5.850 RM -	#5	-4.416	.076 -3.716	.557 -3.0%	2.499	.357 -1.967 .167	*10 - 10 - 10 · 10 · 10 · 10 · 10 · 10 ·	701. 204. 411.	.092511 .101	.050 202 .177	.000040	033 .187 .172	074 .410 .172	-1117	211 Man 701-	
0 - 6.520 AM -	870 HJ NJ 4	1 284 1.900 .195	et	.048 -382 .34	140 996	190 -1-122 .105	.273 -1.996 .195	221.22		9 - 7.700 RM -	5	1.566 1.568 .303	199 1.073	121	****	-076 317	.141	290 -1-447	.502 -2.680	670.	M - 2.00 G - 5.850 AM -	#5	414.4- 608.	.076 -3.716	.557 -3.0%	2.499	.357 -1.967 .167	*10 - 10 - 10 · 10 · 10 · 10 · 10 · 10 ·	701. 204. 411.	.092511 .101	.050 202 .177	.000040	033 .187 .172	074 .410 .172	-1117	211 Man 701-	
0 - 6.520 AM -	ALP CN CN CAS	1 284 1.900 .195	-2-1112155	401. 408. 104. 10.1. 401. 148. 148. 148.	991. 666 041. 01.1	2.16 .190 -1.122 .105	3.24 .273 -1.596 .195	221.2- 966. 16.4	961. 88t 740. EO.	H = 1.10 G = 7.700 RM =	5	-4.31266 1.568 .303	-3.22199 1.073	121127-	.02 .00	716 970. 11.1	2.19 .141	740-1-107- 1766	6.56 .502 -2.680	, , , , , , , , , , , , , , , , , , ,	.3530 M = 2.00 Q = 5.850 AM =	#5	919-4- 508. 48.8	7.74 .676 -3.716	4.63 .557 -3.056	5.54 .455 -2.499	4.45 .357 -1.467	401. 104.1- 402. 15.6 401. 104.1- 401.	1010 BOAC ALLO ASSOCIATED TO THE PARTY OF TH	1010 1000 - 1011 0101		*11. 000 000 II.	41039 .187 .172	211. 014. 440. 10	060. 711 04.1-	6/10 AAGO 7010- 6007-	-4.21341 1.087
0 - 6.520 AM -	CAF CN CN CAS	-4-26 -1264 1-600 -195		961. 906. E70. E0.1. 955. 955. 956. 896. E0.	991. 660 041. 01.1 . 664.	501. 221-1- 061. 91.2	3.24 - 273 - 1.99	221-2- 66. 16.9 19.0	# 10	H 1.10 G . 7.700 RM .	CAF ALP CN	-4.31266 1.568 .303	-3.22199 1.073	1900 1710 - 1717 - 1718 1941	.046.	716 970- 11-11	2.10 .141	740-1-002-06-0	.516	Ava. 200.	•	ALP CN CN	814.4- 208. 48.8	.352 -3.716	450.6- 755. 6.4	5.54 .455 -2.499	741. 740.1- 726. 34.4	+01. +0+.1- +02. Tue.	701	1010 1150- 2000 0101 0560	711, 202 060, 66,	.158 -000 -11. 000 -174	.361033 .187 .172	21. 014. 40 70	060° Lili- 04°1-	CAT	1001
AN - 3810 4 - 90 04 - 91 0:520 AN -	CAS CAF ALP CN CM CAS	190 .4354.26264 1.600 .195	109 - 460 - 11.5 - 11.5 - 170	961" 906" E20"- 60"1" 964" B41" 861"	.174	501. 251-1- 001. 01.5 5.00 -1.122 .105	501. 96.1- 675. 55.6	221.2- 366, 16.4 14.4 15.5	961: 858:- LOO: 80: 120: 61:	H 1.10 G . 7.700 RM .	CAF ALP CN	.321 .53331266 1.568 .303	315 -345 -3.22 -3.22 -3.073	199 121- 91-7-	.347 .962 .0.	716 316 11-11 -076 317	- 1910 -	.335 .521 6.34 .298 .245	342 .516 6.56 6.56 .502 -2.680	A70° 800° 70°	. n oce	CAF ALP CN GN	.2-4 .4-416 B.B4 .805 -4-416	.244 .357	450.6- 786. 6.4 4.6. 955. 955.	999-2- 599- 9996 296- 162-	.226 .351	194 - 194 -	701 664 671 641 641 641	1010 1000 1010 1010 1011	111, 1950, 060, 66, 117,	+210 -900 II. 956 012.	.206 .301 14 187 .172	271. 014. 40 79 406. 702.	0500 6110- 0001- 0000 0000	671° AAB° 791°- 60°7° 966° 917°	.218 .303 -4.21341 1.007
0 - 6.520 AM -	CM CAS CAF ALP CM CM CAS	1.800 .190 .4354.26284 1.600 .195		991° 906° 610° 60°11° 965° 91° 686° 951° 686° 951° 686° 1	1010 110 110 110 1100 1100 1100 1100 1	201- 271-1- 061- 91-2 -1031-122 -105	MAN SANTE MAN STAN INST MAN MAN MAN MAN MAN MAN MAN MAN MAN MAN	27127 986 (4.3) 4.31 .304 -2.122	461. 881. 700. 60. 179. 271. 451. 451.	H 1.10 G . 7.700 RM .	CM CAB CAF ALP CN	1.529 .321 .533 31206 1.508 .303	444 310 447 -3.22 -109 1.073	150° 121° 51°7° 55° 56° 66° 66°	200 -000 -000 -000 -000	716 970. II.II .076. 918. OM	000° 141° 61°C	749-1- 805. At. 4 155. 256. 186-1-	-2.797 .942 .916 6.56 .902 -2.680	Ayo. 200. 70.	. M 3530	CM CAS CAP ALP CN CM	614.4- 208. 48.8 8.84. 805. 46.4-	-3.703 .244 .352 7.74 .076 -3.710	40°C- 756. 64.4 4.65. 955. 41.44.	999.5- 559, 95.6 5.6. 115. 525.5-	-2-007 -2-26 -3-31 -2-35 -2-357 -1-4-4-2-1-4-4-1-4-1-4-1-4-1-4-1-4-1-4-1	000 000 000 000 000 000 000 000 000 00	There appears are on the Court of the Court	101° 115° 200° 01°1 05° 212° 05°-	71. 202- 060- 00- 00- 066- (12- 066-	174 -069 000 II 000 015. 640	172 - 103 - 103 - 104 - 105 - 105 - 105 - 172	211. 010. 470. 70 70 406. 102. 064.	050. 7:1. 04.1. 06.1. 06. 005. 007. 007.	6210 AABO 78101 60070 0060 0170 6501	1.919 .218 .303 -4.21341 1.007
AN - 3810 4 - 90 04 - 91 0:520 AN -	CM CAS CAF ALP CM CM CAS	190 .4354.26264 1.600 .195		991° 906° 610° 60°11° 965° 91° 686° 951° 686° 951° 686° 1	1010 110 110 110 1100 1100 1100 1100 1	201- 271-1- 061- 91-2 -1031-122 -105	MAN SANTE MAN STAN INST MAN MAN MAN MAN MAN MAN MAN MAN MAN MAN	27127 986 (4.3) 4.31 .304 -2.122	461. 881. 700. 60. 179. 271. 451. 451.	G = 7.460 RM = .4220 M = 1.10 G = 7.700 RM = .	CM CAB CAF ALP CN	.321 .53331266 1.568 .303	444 310 447 -3.22 -109 1.073	150° 121° 51°7° 55° 56° 66° 66°	019 .947 .962 .00	716 970. II.II .076. 918. OM	000° 141° 61°C	749-1- 805. At. 4 155. 256. 186-1-	-2.797 .942 .916 6.56 .902 -2.680	Ayo. 200. 70.	. 4.200 AN . 3930	CAS CAP ALP CN CN	614.4- 208. 48.8 8.84. 805. 46.4-	-3.703 .244 .352 7.74 .076 -3.710	40°C- 756. 64.4 4.65. 955. 41.44.	999.5- 559, 95.6 5.6. 115. 525.5-	-2-007 -2-26 -3-1 Let 0-2- 750-2- 1187 -187	000 000 000 000 000 000 000 000 000 00	There appears are on the Court of the Court	101° 115° 200° 01°1 05° 212° 05°-	71. 202- 060- 00- 00- 066- (12- 066-	174 -069 000 II 000 015. 640	172 - 103 - 103 - 104 - 105 - 105 - 105 - 172	211. 010. 470. 70 70 406. 102. 064.	050. 7:1. 04.1. 06.1. 06. 005. 007. 007.	6210 AABO 78101 60070 0060 0170 6501	1.919 .218 .303 -4.21341 1.007
- 9.700 AN - 38.0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	68 CM CAB CAP ALP CM CM CAS	1.800 .190 .4354.26284 1.600 .195	-194 .860 .183 .444 .176 .176	961° 906° 120° 10° 10° 10° 10° 10° 10° 10° 10° 10° 1	130 -130 -178 -178 -450 -160 -160 -160	.216 -1-422 -183 -454 2-16 -190 -1-122 -185	- 11-623 - 189 - 421 3-24 - 278 - 1-596 - 1995	221°2- 666° 16°6   16°6   16°7- 16°7	661° 666° 40° 60° 125° 641° 972° 960°	H 1.10 G . 7.700 RM .	CN CM CAB CAF ALP EN	1.529 .321 .533 31206 1.508 .303	-10 464 315 5263 -3.22 -100 1.073	170° 171° 10°1° 10°1° 166° 060° 060° 060° 060°	.013014 .347 .962 .00	.076316 .554 1.11 .076317	.27 -1-074 -324 -324 -324 -324 -324 -324 -324 -32	740-1- 805. 48-4 125. 126. 186-1- 705.	.507 -2.797 .342 .516 6.56 .502 -2.660	670* 800* 70*	. M 3530	CM CAS CAP ALP CN CM	416.4- 400. 40.8 8.8 8.8 8.6. 4.6. 4.6. 4.6. 4.6. 4.6.	all all all all all all all all all all	450.4. FEL. 444. 455. 455.		Mar. 199-11- 198- 69-9 [FR. 922- 199-2- 296-	STATE THE STATE OF	THE SEASON AND THE COURT SEASON CONTRACTOR OF THE CONTRACTOR OF TH	TOS 16.00 PRO PRO PRO PRO PRO PRO PRO PRO PRO PRO	74. C. S. C. S. C. C. C. C. C. C. C. C. C. C. C. C. C.	.011 -000 -1000 11000 -0000 -114	201. 181. 600. 10. 10. 102. 671. 920.	211. 014. 470. 70. 44.	0600 Alle 00011 2000 0020 0020 02101	Marie   Ma	-340 1.919 .218 .303 -4.21 -341 1.007

Table 43. Aerodynamic Coefficients for Configuration BIT13X1

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* *	AT A	-4.27	6.	1::	3.27	4.35	-2.11	•		ALP	-4.38	-3.26	-1.00	50.	7.27	3.30	9.4	\$0.	•	ALP.	35.	57.4	\$ . 28	4.20	*:-	5.09	1.52	9.7			01:1-	-1.63	-2.16	4.20
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	CA8	22:	~ ? ?	<b>:</b>		:	.113	;	· Z	CAS	.172	3.	70.5		3	601.	.172		ž	CAB	74:		3			.127	• 125	?	62:	?:	. 121	:21	•123	.12
4.7%	5		-018	12:	200	1.343	.033		•	5	941	8	125	3	- 200		1.241	.2.287 .046	4.930	5	-3.249			701		570	412	- 286	-135	999	244	114.	.532	1.152
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	AL.	-3.18	 	01.1	2.18		9.2. •			ALP	4.34	-3.24	***	50.	*1.1	7.74	*	 .0.		4.6					3.26	2.10	1.63	.0	Š	ė.			-2.08	• · · · · · · · · · · · · · · · · · · ·
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• 4.520		1-145 -102 -791 -0095 -233 -004							. 7.700 4%	_	_		_					02. 100	. 5.890 .		-3.115 .17										• • • •	***		
.00 0 . 6.520	5		010	7		-1-606			*10 0 * 7,700 AN	5	1 1 1	1	125.	420	•		2		8 068.8 . 0 00.1	*5		2.20	-2.018				3		*:	017	_		. 553	1.201
925.0 . 0 . 0 . m	5	 		10.		.321 -1.400	237 1.143		m . 1.10 0 . 7.700 AM	<b>85</b>	1.163	200	126. 011	460 410-	• • • • • • • • • • • • • • • • • • • •	641.	962-1- 982-	-2.001	A 2.00 0 . 5.890 A	5	-3.119	***************************************	B10-2- 200.	100-11- 216-	234	200	100	.072 247	.034154	110 600*	021	100	132 .993	.204 .809
	ALP CN CM	-4.29 -239 1-145 -3.10 -160 -701 -2.10075 -233	0.1.0.0 %00.1.0.0.0.0.0.0.0.0.0.0.0.0.0.0.0.0.0.	1.10 .004	3.24 - 42. 42.123	4.32 .321 -1.600	-6-29237 1-149			ALP CN CM	841-1 746-4 18-4-	-3.22170 .040	126. 011 +1.5-	420 - 410 40 - 420	111 040 1111	E 00 - 001 - 01 - 0	967:1- 987:	.03 .011002	•	5	\$111.6- abo. 18.0	7.70 .564 -2.506	B10°Z= Z04° 00°4	200.1- 576. 00.6	\$00-1 412- FE-E	2020 - 2010 - 2020	1.70 .100 445	1.16 .072297	-13e	110°- 500° 60°	7.65027	100°1 00°1	-2.06132 .953	-9.19204004
025*9 + 0 00° + H clas* + SH	ALP CN CM	.413 -4.25239 1.145 .419 -3.18 -701 .424 -2.10075 -233	0.00	975- 980- 01-1	21.1. 44.5. 42.6	.392 .321 -1.606			· # 027	ON CHAPTER CM	1845 TAC. A 18. A.	. 50A - 3.22 - 178 . 046	125. 011. +1.5.		1011 004		960°1" 980" 66"* ee*	.901 011 005 .901 011 002		247 CN CN	\$11.6- 000. 10.0	90.5- 496. 01.7	B10°Z= Z@4° O@*@	100°11 216° 06°6	100° 410° 80° 100°	2000 - 241 - 422	. 310 1.70 .106445	795 270. 01.1	. 116	710°- 600° 60°	1313 - 24.	100*1 60*1 075 *	.313 -2.06132 .553	946 408" 61"6" 91"6" 916" 108" 68" 108" 108"
C186" - ## C	CAB CAF ALP CN CM	.122 .013 -0.29 0.230 1.105 .120 .010 -3.10 -1.60 .701 .110 .020 -2.10075 .233		414 .429 U.1 1.10 .004	100	-129 .392 4.32 -1.600	111 121 121 121 121 121 121 121 121 121		• H 027** • NH 0	CAR CAP ALP CN CM	200 - 10 - 10 - 10 - 10 - 10 - 10 - 10 -	.204 . 204 . 3.22 178	126. 011. 41.5. 514. 504.	460° 460° 40° 40° 40° 40° 40° 40° 40° 40° 40° 4	.204 .407 1.11 .079319	202 - 470 - 210 - 202	900" 900" AC"+ 400" 400" 600"	.215 .483 6.36 .447 .2.081002	•	5	\$11.6- 600. 18.0 6.2. a15.	906-2- 496. 01.7	B10°Z+ 200° 00°0	100-11- 216- OK-6	#00-1 107: Park 107: 401:	200- 201 62-2 (10 64)	179 010 1.70 1.70	179 .912 .116 .072297	961. 960. 20.	710- 500, 60,	.172 .319027	1901 - 201 - 201 - 201	-175 -313 -2.06 -132 -553	980 400° 51°0 - 916° 671° 671° 671° 671° 671°
	CA CAS CAF ALP CA CB	18 1 12 1 12 1 1 1 1 1 1 1 1 1 1 1 1 1 1	070 620- 0011 000 010 011 001 010 010 010 010	48 400 OI'I 620 0II' 400'-	621°1° 462° 42°6 114° 411° 416°1°		-1806 117 -417 -418 -429 -421 -2189 -4219 -4189 -4219		# 7,440 BR # 2420	Cas cas als als	200 - 200 -	.053 .205 .506 -3.22 -170 .044	126. 011. 41.5. 514. 904. 754.	##2" ##0" #C"	- 11. 10. 10. 10. 10. 10. 10. 10. 10. 10.		## ## ## ## ## ## ## ## ## ## ## ## ##	-2,270 .215 .483 6.56 .447 ~2,081 .5.08 .533 .207 .401 .002	* * COS6* * NA . 37** * P	HO CAR CAP GAS ALP	\$11.6- 400. 18.0 6.05. 615. 405.6-	905-2- 995. 01.7	BED-2- 200 00-0	LOGGE PLACE OF CO. C. C. C. C. C. C. C. C. C. C. C. C. C.		100° 00° 00° 00° 00° 00° 00° 00° 00° 00°	244 - 401 010 1.70 1.00 . 104	792 - 270 . 01.1 516 . 471. 016	961. 860. 50. 10. 116. 141. 101.	710°- 500° 60° 215° 711° 420°-	.130 .172 .313 65	-284 -171 -340284 -403 -172 -317404	666. 561 40.5- 616. 671. 476.	400° 400° 410° 410° 410° 400° 400° 400°
C186" - ## C	CAL CAB CAF ALL ALL CR	.122 .013 -0.29 0.230 1.105 .120 .010 -3.10 -1.60 .701 .110 .020 -2.10075 .233	070. 250. 40.1. 050. 411. 451. 460.1	411. 400. 01.1 (429 11.10 000. 11.1	62(2) 46(4) 47(4) (14) 41(4) 41(4) 46(4) 46(4) 46(4)	. 147 -1.89 . 129 . 302 4.82 . 321 -1.600	.007 -20374 -124 -124 -124 -23 -232 -232 -232 -232 -232 -232 -23		• H 027** • NH 0	CN CM CAM (AF CN CM	241 TACA 16.44 (642)	000 001 0010 000 001 0010 0010 0010 00	126. 011. 41.5- 512. 512. 511.	##2° ##0°° #0°°°   60°°	# # # # # # # # # # # # # # # # # # #	600 01°C 01°C 010°C  ## ## ## ## ## ## ## ## ## ## ## ## ##	.215 .483 6.36 .447 .2.081002	•	SA CAR CAP ALP CV	\$11.6- 600. 18.0 6.2. a15.	- 24. 72. 72. 72. 72. 72. 72. 72. 72. 72. 72	BROOM NOOF DOOR EPUT FOR STREET NATURE	TORRES OF THE TORREST OF THE TORREST PARTY TORREST TOR	100 1 410 200 100 100 100 100 100 100 100 100 1	500° 40° 52° CIC 61° 61° 61°	540 - 401 001 016 641 660 401		.030:01 .173 .311 .02 .036150	710 600° 40° 211° 211° 420- 600°		190°1 90°1 076° 121° 90°° 460°1 (90°°)	666. 561 40.5. 618. 671. 476. 461	-200 1020 1020 1170 1150 1150 1201 1201 1201 1201 1201	

Table 44. Aerodynamic Coefficients for Configuration BIT13X2

.41%	CAF	:			422	7	ş	.367	.37	. 363	362	.397			.3860	CAF		376	900	200				300	2	.356	.385		.3100	CAF	-235	. 229	. 221	*22.	***		234	,	. 239	.241	.241	.239	2.00	200	.235	
ž.	CAB	į	162	2.2	280	.281	ž.	٤.	8	.315	.323	230			· Z	843		8	96:	26.				5 5	.214	.223	-18		*	5	.115	-11.	9:	•	1	::		901	901	100	108	901	907	.00	10	
1.090	5	;	82:	7.4	.082	157	401	797	-1-160	-1.630	-2.741	•			. 8.560	ž	•	1.613	. 189		714.			196	-1.677	-2,976	012		• 4.130	5	-3.949	-3.086	-2.410	-1.859		010.1		A18.	-161	.020	991.	.332		000	1.43	
1.00	3	;	240	***	025	.032	•	.163	•534	•314	.503				1.50 0	2		304	221			3		101	368	.569	.016		3.00 0	3	408	.650	• • • • • •	504			2	1,00	98	003	037	40.		22.	315	
	ALP	,	17.4-		10.1-	Š.	1:10	2.18	3.26	4.34	15.9	1.10				41.6	•	-4.38	-3.27	7.7-		2 -		77.7	. *	6.73	ş			ALP	6.55	7.67	6.37	5.29			55	60-1	.50	10	55	-1.09	-1-00	AL	4.25	
0604.	CAF	;	.345		9	380	.366	.363	.353	154	.345	.363			.4160	347	į	. 420	.432	. 433	.430		999		514	392	.437		.3100	CAF	.253	.245	•5•0	.243	2	367	256	259	.262	192.	•52•	.261	262	256	.254	
ž	843	;	95:	2		.227	. 225	.231	ş.	.247	•52•	. 558			z z	•	5	•12.	.213	210	602	707	.210	027	376	2	506		*	840	.152	.153	.152	•				7.1	136	137	.137		2 2			
. 6.750	5	•	1.495			226	472	983	-1.410	-1.179	-2.901	229			400		;	1.523	1.061	707	9	770	- 365			-2.686	000		4.930	5	-3.629	-2.903	-2.340	-1.664	7	160.1-	42.5		180	015	.328	•				
. 95	3	. ;	270			•036	10.	.180	.256	.338	.515	•036			1.30 0	į	5	285	- 200	127	- 0 2	20	60.			525	510	:	2.50 0	5	.755	91 9.	- 205	.03		057	7		040	8	070	101		4000	302	
	ALP	. ;	4.26			6	1:10	2.18	3.25	4.32	6.49	•03			=	•	ì	-4.35	-3.24	-2.15	90.1-	\$	= :	2.43	3.33	4.4	å	•	*	AL.	6.73	7.60	6.50	5.42		3.27	94	=	.5	00	-1.02	-1.55	-2.07	1	17:5	
9	, ,	5	.375	.375	.374	976		147	1961	350	347	176	.379		4230	317	<u>.</u>	694.	76	P. 4.	٠,	199				0.4	194.	**	3500	CAF	.279	*282	-585	.260	***	*27	200	***	285	.207	.207	.284	982	***	.205	
, ,		•	157	.145	137	<u> </u>				20	000	138				:	6	.276	.270	.267	902	5.7.0 	273	276		ž	273	*7.	:	C.A.B.	161	6.	189	• 100	.185	Ξ:	2	7.	112	172	.172		*:	2:		:
6.520	_	5													7.700	į	5	1.386	96.	. 597	.255		60	747	170	ğ	300		5.850	5	446	046-2	6.4.2	1.567	.580	9	26.			900	181	.343	250	5	6 6	
0 06	<b>.</b>	5		_	_		_		•						1.10 0.1		5												2.00 0	_	·	_	Ċ	_	•		_		_			_	_		500	
			-4.25	-3.17	-2.11	60.7-						00-1	60.			•	ì	-4.30	-3.21	-2.13	-1.05	•	7:	5.19	3.28		37	1:1		41.8	47.74	7.0	•:30	5.47		2.30	5.23			60	•	-1.02	-1.55	90.2-	-3-1-	•
0106.	<b>₹₹</b> 5	92.	\$65	661	~	•		2.5		*		4	154	345	47.50		, <b>4</b> 7			47.4					436	25	•		9900	345	•	•0	Ď,	* 01						-	13	91	•			
· .															•		CAS								300								Ī		•		•			•		Ī		•		
5.700 H	(A) (A)														98 004				•										.200 84	.m. C48															200	
•		604-1							•	•	,	•	•		•	•	7								23 -1.092	•	•		6.70		•	•	•	•	•											
•	5	·			_							_			60.1	•			•				_		223	_	_	_	:													•				
	à	*	7.7		· · ·	3	-							;	•		4			,	**	•	• .	•		;	•	:	•	1		7.7	6	<b>?</b>	•			•	•		•	?			7	

Table 45. Aerodynamic Coefficients for Configuration BIT13X5

9		5	.397	5	Š		150	200	‡.	13.		.3860	;	Š	.377	.395	3.00	•	9	.397	.397	9	.385		.3100	CAF	:	26.7	22	.227	.231	.227	-232	. 53.	244	.231	. 23	.235	.230	.233	230
2	: 3			.276	÷.	707	717	.319	.324	.334		=	;	Š	.227	.22	7	117	.217	.221	525		.232		*	<b>64</b>	:	77.		172	.122	77	2:	22:	171	122	122	122	122	122	122
7.090		5 ;	.952	919.	47.	424		1.339	1.793	-2.892		9.560	;	5	. 109	1.277			-510	***	437	. 986	1.903		4.130	5	:		3	. 192	.426	-021	.635		501	200	.179	š	.537	6693	1.482
90.1										Š Ž		9	į	5	309						•	•	•		9	3					_	_						_			316
		1	7.7	-2.12	50.7-	3 5	27.7	3.25	4.32	2.17		# # 1.50			-4.37					_			_		M . 3.00	A.C.										•	Ċ	·	•		4.33
.4340	Ç <b>≜</b> F		0	116.	040	34.	53.3	.314	4312	.35#	.4160	CAE	j,	9		524	.423	.425	•		.391	. 393	215	+24.	9100	CAF	.253	067	.253	.247	9.7.	267	251	.255	.255	.256	.253	.253	<b>.</b> :	.262	.259
÷	CAB	546	.222	2	200	.213	•231	. So	6524		=	3	•	192.	62.	737	.231	.234	į		.270	75	.235	.228		843	651.	*	.158	.157			1	3		.147	÷:	947.	2	3	141
05.150	ð	1.4.28	2.5		7117		947.	.403	100	. 523	9.400		•	1.733		9	-,030	7			2.933	2.914	~~	-,020	4.430	5	_							_	_	_	_	_	2.5		_
		-282						•	•	•	1.30												3:		•	3	Ĭ.		•	٠.				_					1 5		
•		****										•					_						);;;	_	N = 2.50	ALP											•	•			•
	***	316	.335		*	.333	.33			i.		06.24-	<b>50</b>	į	3	•		3	.45	•63.		.397	į		3500	**	.270	5120	.271	.276	107	. 205	982	.267	.200	.288	**	.287	-247	26.	<b>4</b> .
į	3	•	<u> </u>	15	557	.193	. 155	•		15		:	C.A.B.	į	Ē.	ê		ž	ŠÉ	Ę		.312	ī.			3	212	.210	Ž.	ē:			102	201	= =	=	9	2:		=	<u>:</u>
2		7	•		•																				2	_								-		-		-			
***	5		27				-1.09C			9		7.70	3	•	£:			-015	- 304	20.		2.473	3		:	5	-3.800	-3.130		-2.027			***	370	197	013		Ş		1.13	-
	5		27		067	572	•	. '				.10 6 - 7.70	83		-257 1-28		_				٠	_			9.6 . 9.00	5	•			•			_		_		_		907.		
26.4 . 9 . 4 8	#3 #3	1.031	77.7 el? -			1 .101 572	•			2		M = 1.10 G = 7.70	5					.00	.072		Ž	•	į		R - 2.00 Q - 5.8	ארם כא כא	**	=	•	9		182		•0.0	.039	-005	033	072	101	226	304
•	ALP CN CN	160-1 016 52-4-	250-1 012 01-0- 250- 001 11-0-	-1.00	051 650 150	1.07 .101572	2.10	962° 67°6		201.		-	#7 ATE		157.	23.6	240-1-40-1-	.02 .067	1.10 .072	2-10 -137	4.35	6.63	200		•	3	4.75 .736	110. 69.2	***	104.	000 00 H	24.19	4.	1.10 .076	.57 .039	200* 60*	660*- 64*-	-1.04072	-2-11140	-3.10226	-4.27309
	#3 #3	160-1 016 52-4-	77.7 el? -	-1.00	051 650 150	1.07 .101572	2.10	962° 67°6		201.		4230 m 1.10 G 1.70	5			23.6	240-1-40-1-	.02 .067	1.10 .072	2-10 -137	4.35	6.63	200		8.3 ° 0 ° 5.00 ° 0 ° 5.8	_	4.75 .736	110. 69.7	***	104.	000 00 H	24.19	4.	1.10 .076	.57 .039	200* 60*	660*- 64*-	-1.04072	-2-11140	-3.10226	-4.27309
	ALP CN CN	1890 016" 65"0"	250-1 012 01-0- 250- 001 11-0-	27. 040 04.1.	061- 000 20- 180	245- 1917 215	. 191. 91.2	262 576	202	100 100		1 - 4	#7 ATE		157.	6810- 2765- 9600	\$40.1 40.1.	100. 50.	1.10 .012	2.10 .137	- 195 4.35 - 591	66.0	000 20.		. n . 3300	3	.313 8.75 .736	110. 69.7	**** 26.4	104. 44.4	11.0	.312 2.15	.317	.911	.313 .039	200. 60.	EEG*- 04*-	.311	-2-11140	-314226	.313 -4.27309
•	ALP CN CN	1800 016 - 52.00 016. 605.	- 321	occ. 640 40.1.	061 020- 20-	245- 100 -001	. 101. 01.2 	202 000	200 200	100 . 302 1.00		7.460 AN4220 H - 1	1 CAF ALP CR		167. 06.4.	110 At 5 - 464 110		. 312	510 01:1	200 120 137	. 10% . 26.4	699 66.0	900. 20.		. 13000 n .	CAB CAF ALP	461, 213, 8,78	110. 69.7	56.0 DOC 05.7	1000 0000 0000 0000 0000	C10	.270 .312 2.19 .152	.210 .317 1.04 113	.216 .311 1.10 .076	.218 .313	200, 60,	ERO*- 010	210 - 110 - 1100 - 1200	113 -114	.222 .314226	.229 .31927309
4 9.700 AM5010	NO NO 674 640 640 NO	1800 016 - 52.00 016. 605.	121 - 121 -	27. 000 - 00°1.	061 20- 20-	245- 101, 90-1		262 262 262	2.978 .290 .202 .203	957 - 100 - 302   1-00 - 102		0 - 7.440 AN4230 H - 1	1 CAF ALP CR		162. 06.4- 110. 414. 111.	100 - 77 C - 960 116 190 C			1.10 .012	1.050 .336 .467	. 105. 26.4	600 66.0	980* 20*		- H - 1,700 AM - 13500	CAB CAF ALP	. 154 . 154 . 313 8.75 199.t-	110. 60.7	661 2619 BOK ON 1997	1040 44.4		91. 21. 21. 2.19 152	401 317 1.04.	11 116. 015. II-10 .076		200, 60, 116, 911, .09,	- 107 - 04 010 -	. 340 40.1- 11.0 - 1.20	.766 .276 .313 .2.169 .2.469	1.192 .222 .314 -3.19226	1.000 .229 .319 .4.27 -4.300
	CH CH CAS CAP ALP CH	1693 0200 - 52.00 020 5000 10001	201 012" 01" 1" 1" 1" 1" 1" 1" 1" 1" 1" 1" 1" 1" 1	1052 201 100 100 100 100 100 100 100 100 10	061- 20- 20- 116 011- 20- 220-	240- 101- 001	191 91 2	202 202 202 202 202 202	200 200 200 200 200	100 -100 100 100 100 1000 1000		. 7.46 Am4220 m . 1	CR CAB CAF ALP CR		667 - 6679   160 - 675   160   160	100 - 77 - 460 - 110 - 410 - 110 - 110 - 1	540. 40	700. 50 515 700.	10. 01.1 100 10.10 10.10 10.10	751. 01.5	. 102. 205. 206. 416.1- 402.	690 66.0	990° 20°		* * * 200 RM * . 3500	CN CN CAB CAF ALP	467. 27.4 FIE. 121. 144.6- 447.	110. 60.7 20.00 20.00 20.00 20.00	ARRY PARTY BOAT OFFICE SPRING PORTY	10% 44.4 80% 40% 40% 811.57 80%	200 000 000 000 000 000 000 000 000 000	251. 61.5 215. 61.5	.110001 .210 .317   1.04	.070. 01.1 1.10 .015. II-10 .070.	***************************************	200, 60, 116, 812, 700, 100,	600- 64- 030 030 41- 40-	20°- 40°1- 118° 812° 846° 870°-	.766 .276 .313 .2.169 .2.469 .2.469	-231 1.192 .222 .314 -3.19226	113 [.600 .229 .3134.27300

Table 46. Aerodynamic Coefficients for Configuration BlT13RX1

N = 1.00 G = 7.090 RM = .4190	5 N	-4.25218 1.042 .172 .422	066 .390 .130	501. 261. 160-	***************************************	111. 657 600. I	051 078 - 001 -	1261 -1-226	107 -040 -2.401	171. 1.010 .171		H 1.50 0 * 0.560 RM * .3860	ALP CN GH CAB CAF	110037 .211	053235 .202	015294 .193	200-	011. 057. 010	2020 950 0200	146032 -224	364 -1.093 .236	.021 .467 .203	B/1: 5+2: 510:-																		
M = .95 G = 6.750 RM = .4090	CN CN CAB	524. 610. 850. 691 25.4-	-113 .368 .084	450° 650° 610°-	100 000	.052112 .072	.046 259 .093	111. 99 671.	.204 739 .122	.391 -1.760 -140	***************************************	M = 1,30 Q = 8,460 RM = .4140		-234	041 .073223	037057 .210	017 .021 .188	015 .254 .189		037 162 160	164 252 - 241	.362 -1.351 .255	016 -255 -109																		
0104. * ## 058.6 * 0 08. * #	SAC CN	400 Miles 1997	-105 .361	05	700 900 PTO-	200 -117 -102	.007 700	201	.204 745	.349 -1.494 .172	660° 961° 650°°	H - 1.10 6 - 7.700 AM4230	ALP CN CN CAB CAF		****	962* BBE* BET**	041 204 204	.000 .071 .204	.040 049	.067212 .233	662. 084 441.	782 - 1-804	.001 .062 .211	M = 2.00 G = 5.850 RM = .3500	ALP CN CN CAS CAF	.540 -2.152 .202	-421 -1-367	002. 161. 162.	401	191	454. 400	**************************************	024 174 174	6.1.	D. 161. 010	47.	271. B4. 410.	.004 407	100	101	
#	ALP CN CN CAB CAB	6017 6297 12177	062. 851. 844. 411. 51.E.	27. 47. 900	211. 000 020-	101	ALL - 104 701	196 - 196	200 - 807	. 343 -1.473 .180	.044250 -121	n - 1.05 G = 7.460 RN4220	476 CN CB CAB	. ;	***************************************		200 - 11 - 100 - 1	.00035	. 094 203 -194	.104 401	192. 961.	922 925 C 977	620 660 4120 620	# 1.75 G * 8.200 AN # .3500	ALP CN CN CAS CAF	1930 -2-010 -241		.20630	**** *** **** ****	7.57 .051 .454 .254 .304	.000	004 .554 .210	017. 644. 410.	+02° £9K. 910°-	00-	200- 600-	100° - 17° - 100°	416 167 - 610 -	*13. 136. 010°-	22. 0.5	

Table 47. Aerodynamic Coefficients for Configuration BlT13RX2

.4190	CAF	462	1447	.3860 CAF	44444444444444444444444444444444444444		
, Z	CAB	55 59 123 123 123	225.58	RN CAB	224 225 202 186 186 208 208 233 248 194		
7.090	5	1.575 1.045 .591 .196		8.560	. 043 - 153 - 118 - 118 - 118 - 118 - 206 065 - 1123 - 103		
1.00		109 109 039		1.50 02.1			
E	410	-3.26 -3.18 -2.11 -2.11	2.12 2.13 2.30 2.40 2.40 2.40 2.40	_	12.02 12.03 12.03 12.03 13.03		
060*	CAF	;;;;;;	25. 25. 25. 25. 25. 25. 25. 25. 25. 25.	.4160 CAF	.397 .650 .610 .612 .603 .603 .611		
=	CAB	1,000,000	0.095	# 45 CAB	.234 .2237 .2238 .223 .223 .244 .264 .264		
6.750	5	1.317 	-672 -696 -1.415 -2.451 043				
÷.	5	100	198 198 198 198 198 198 198	1.30 GA			
	ALP	77.17	12.2.4		1.28 1.004 1.004 1.004 1.004 1.004 1.004 1.004 1.004		
010	CAF	4000 4000 4000 4000 4000 4000 4000 400	. 3 90 . 3 90 . 3 90	.4230 CAF	\$	. 5 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	
=	CAB	60000	0000	. 442 CAB	25 243 222 223 224 224 224 224 234 234 234	E	
0.520	5			. 7.780	15.30 28.3		
•	5		20. 00. 171. 286. 485. 181.	_	1000 1000 1000 1000 1000 1000 1000 100	2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2 2	03
:	A.L.	-3.14	1.00 2.12 3.18 4.24 6.37 -4.20		10.20 10.20		5.4
.3410	CAF	282	200 200 200 200 200 200 200	-4220 CAF		§ 3 <b>!</b> !!!!!!!!!!!!	2
:	87	00.000	100	* 64 0	2236 2236 2236 2236 2236 2236 2236 2236	22.23 B B 8.22.23	.23
5.700	*	1000		3 .	10.01 10.01 10.01 10.00	- 200 - 200	* 35
	5	0000	•	ā	12.200000000000000000000000000000000000	44 7 7 4 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7	_
	*	20.7 20.1 20.1	22.22.4		7777		

Aerodynamic Coefficients for Configuration BlT13RX5 Table 48.

96 14	S.	414	- 4.5		.428	. 428	914	9	203	2		. 366	ÇA	.413	9 5	3	303	2	.399	Ş.	Š		391		.3100	CAF	286	.281	20.	.281		416	325	330	.345	Ž.			316	.317	Š	
Z	CAB	230	77.	2.5	2	591	802	232	2.5	•	i	r K	CA8	+12.	25.5			2	2	.213	<b>*22</b> •	900	194			<b>8</b>	0,60	3	90.	8	66		200	0.0	•00•	200	6	3	9	1200	.031	
7.090	5	46	9		7	•	8	*	52	•	;	260	5	241	92	334		22	3	582	320	. 5	113		4.130	5			1.5	\$20	2	7		:	į	7	ī		3	3	233	
•		1.064				·	٠	•	•			•	_	_	٠.				_	_								٠.														
1.00	5	301	128	200	60	.180	.257	*	.561			.50	5	60	-05	-	5 6	3 8	6	.05	77	35		:	9.0	5	;	į	.33	.21	~			0	02	ě	Ö	6	50	.0	-	
	ALP	-4.26	-2-11	<u>.</u>	. 6	2.16	3.24	4.31	6.49	5.16		*	AL &	4.30	-3.21	-2.12	8:	9	2.19	3.28	4.3	70.0	2 6			AL.			6.3	5.30	*:5	3-17		4	5	ē	•	8:		-	=	
			•	•																																						
																																									_	
060	CAF	427	1	5		1	43	435	ě.	.427		.4160	CAF	.381	386	.308	.36			.387	.385	.386			.3100	C.A.		.320	9	.314	.333	~		356	365	.369	100	.362	345	*	33	
=	842	490	\$	£20.	0	5 9 0	3	700	141	•		z	<b>648</b>	23.6	.225	*17	2	ē	217	162:	.241	.263	ž.		=	<b>8</b>	;	66		.07	6	3	700	250	3	\$40		9	6	9		
6.750	5	200	*	\$	<b>.</b>	2	```			•		9.400	5	5		8	8	=	82.	2.0	1	<b>1</b> 24	131		4.430	5	;	<u> </u>	2	2.	•	į			5	53	5.5	3	7	77.	: 2	ľ
•		1.637	•	•	? ;			-	~	:												٠						-2.054													_	
÷	5	472.	. 128	05		:			1	27		1.30	5	4	8	5	0	ŝ	200		-		Ş		2.50	5			~	5	ş		2			8	ģ	.0	9	5	9	:
	ş	\$2.4	2.11	8.7	ģ.	5	2.22			4.29			¥	:	-3.20	-2.12	-1.04	9	2:			. 5	.03			A.		11		5.36	4.27	3.17	::			9	*	:	-1.51	5.5		;
		•	•	•											•	-																										
0100	4	355	372	.378						:		.4230	3	;		•	.42	7.	429		9		. 425		3500	3		. 36.			.30	.374	*				*	•	.36			:
:	GA B	*20.	8 8	010	8		250		2 6			=	6 A D	į		522	210	2	112		256	ê	.218		=	5		3:			132	*	÷	2	6		2	2	2:	2	ž:	
920												7.100		,	2 2	: 8		2	Ď.	5.4	; ;	. 2	1		\$. <b>8</b>	5	,	;			5	5	£	•	2		12	Į	=	È	<u>.</u>	4
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٥	5	129		.03	.0.	=	6	707		:		01.1	5	. '			d	0	8	-			2		2.00	3	•	*		7	00	Š		8			200		90	90.	\$	5
:		-3.15	•	?		:	2						4					ò	. 10	2.17			2.1.2		•		•	:				2.5	5:19	3.	•	ċ				-2.05	3.0	2
•		ï	ĩ,			. •		•		•					•	' '	•																					•	•	•	•	•
30.00	3			. 207	~	~						270	CAF	414		* 74.		~		424	=	<del>-</del>	?		9		į	. 384	. 365			9		.378	.376	.37			300	.367	.372	.37
:		0.00		1	~*0	2	6						5	374	2.0	23.7	572	=	553	2	2.0	\$	.223			:	•	2	ę	ć:			2	•	:	9				:		<u>:</u>
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a	5	161:-		660	.0.7	.077	e .		22:			÷	3	•		132	.00	.0.	.00			*	.00		•	• ;	,	*7.	.35	??	-			.031	.027	-1017	800			200	10	027
		·				_		_	_			. 1.05			. ,	•	٠						_			•			_		_				_			_			_	_
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Table 49. Aerodvnamic Coefficients for Configuration B1T13SX1

AN4190		.177										RW = .3860	CAB CAF									150 .392		3N * .3100	CAB CAF	A55. 801.		_					•099	•			•	
7.090		1.657										3.560		,		Ĭ	•	•		•	•	251	•	• 130	5	-1.710												
1.00		299							•	•		. 50 0	3							٠	•	•		3.00 0	5	. 783	•		•	•								
	¥C.	-4.30	-3.22	90-1-	.09	1:10	2.18	3.26	¥.,	6.53	.03	•	- ATA	1	-3.27	-2.17	-1.05	\$ -	2.25	3.37	*		-1.06	x	ALP	9.40	7.36	6.27	61.6		2.01	1.45	-92		9.	-1.19	-1.70	
90	<b>.</b> 5	.453	094		854.	3	2445	.432	.426	• 450	£\$4.	. 1 60	CAF	145	194.	994.		- 1	694	.463	*	994.	.467	.3100	CAF.	742.	.247	.242	-244	7.7.	252	.247	#*Z*	2.5	952	.248	.247	
2		.120	=:		.122	.122	.122	.126	.131	3	121.	2	0 <b>4 8</b>	169	9	.164	<b>=</b> :	3 3	191	.165	7	3	.161	=	Q V	3	3	.130	-138		124	121	<u> </u>	2	2	21.	.120	
. 6.750	5	1.636	1.479		015	7.347	705	-1.001	-1.313	-2.245	.351	B.400	5	1.37.	.965	. 585	.257		04.	-1.153	-1.572	362	. 268	6.430	5	-3.521	-2.793	\$12.5-	-1.749	7.5	035	469	: ·	0	.155	.317	Į	
\$	5	316	***	120	210-	000	.152	.217	*87	.45	03	1.30	Š	272	191	-: 11	400-	200	.171	*52	.337	.00	045	2.50	ð	*	*! •:	.497	. 39	220	-	•10	20.	000	036	073	- 108	
	AL.	67.4	-3.21	7	200	1:10	2.17	3.25	4.32	6.50	-1.09		ALP	-4.35	-3.25	-2.15	-1-03	6.1	2.25	3.36		::	-1.05		ALP	19.67	7.56	4.0	9.3	1.22	2.15	1.59	1.06	100	53	-1.07	19.7	
0104.	3	.395	÷.	9	9	*.	.392	-385	.382	•10		.4230	CAF	.510	.536			. 536	. 527	.529		.534		.3500	CAF	.276	• 5 2 9	082	200	.276	-274	**	273	.276	.279	1		
2	3	` <b>6</b>	ŝ			<b>~</b>	60.	•	*!			2	43		5	::	2	=	.13	=:				:	CAB	.175	2	3 3		8	\$	•	1	.142	745	?:		•
. 6.520	5	1.50	2	9	191	115	455	*	-2.901	146.		7.700	5	1.410	7		960-	1		-1-147	-2.505	.78		5.6%	5	-3.320	-2.727	10.		- 90			*	002	:			į
?	5	201		000	00	.056	.122	202	. 525	B60.		0 01.	3	277	061		105	ē		2	90	-139		0002	5													
*	7	17.27	-7.7	50:1-	6.	5·-	***	3.23	**	-1.09			46	-4.33	-3.23	-	6	===	2.20	3.29	25.4	-2.19		*	41.4	2.7			•6.	3.20	2:	9	*	<b>7</b>				***
.3410	CAF	***	~	.437	•							. 4270	<b>¥</b> 5	.510	•25.	0.5	425	***	.524	. 526	Ž.	.531		• 3 300	•	£.	* .		500	30	ž.		267	\$ 7	2	8.5		
:	5	::	:	:	=	:				::		:	5	.100	•	ė	Ę	.192	.10	161	7	é		:	5	\$0.	ê:			=	ž ;		~	-	2 :	:		
9.790	5	1.31	9	.235	-:13				2.3.3	*		7.40	5	1.272	1.171				-1.013	-1-316		*		• • .700	5	3.436	-2.61		1.374	-1.002	6	2	2	•050	*			
•	ŝ	*62	•	***	200					*20	•	1.09 0	5	262		151	90	150	- 512	*	į	.000		1.73 0.1	5				•	•								
	1	67.	::	\$	?			3		70.		•	1		3.23		į	:::	3.27	<b>6</b>		.0.		•	;	:			:	.30	~-	:	į	=;		3	10.	

Table 50. Aerodynamic Coefficients for Configuration BlT13SX2

0414.	Š	7		.422	.415	ş	\$	¥E.	.362	-417	;		CAF	747	ž	.391	.399	ş	9		.363	.368	3 :	166.		3100	CAF	9	,;;		2		220	737	230	523	245	.23	. 231	.234	.233	.236	. 235	.22
į	3	82	27.5	. Z	1304	. 8	ýģ.	.314	.338	.301		E	CAB	212	202	.195	.192	2		è	1120	*27*	£:	<u>;</u>		z z	84	:	===	::	::	:=	::	0	0	0	100	108	100	100	.104	-104	•01.	• 10
7.000	5	1.376	200	.132	457	40	-1-179	1.50	.2.641	+10	9	006.8	5	2.588	1.067	<b>6</b>	.277	000	00.	1.270	1.025	3.105	562.	e v		÷.	5	***		21.5			110	199	203	324	4	240	512	**	.577	738	1.110	1.513
8		229					-		•		9		3	-,303		_	_									8	5			'								_		_		_	_	
	ALP	-3.21	-1.06	.00	01.10	71.2	3.25	4.32	4.50	1.04		•	ALF	-4.34	-3.27	-2.16	-1. 8.1.	8	• • •	7.	4.47	6.73	2.5				ALP	;				91.4		20.2	74-1	3	3	60.	7	-1-16	7.7	-2.21	-3.27	4.4
0 <del>,</del> 04.	CAF	204.		\$65	-405	ŝ	2	2	2	**	917	3	CAF	.422	9	*	*			436	.43	.432		i.	;	.3100	CAF	356	747	546	240	24.5	. 253	2,20	.252	*	. 259	982.	962.	-259	.255	797	957.	9. 7.
=	CAB	8.	215	.218	-217	210	.217	252		247			5	.251	**	•532	.22	•219		2.7	.243	₹.	2.5	23.	į	2	5	91	5	3	1	?	2	13	-136	=	.135	.135	.136	.134	.134	.137	361.	*
. 4.750	5	46.1	. 070	.347	-113				24.	-3.340	4.400		5	1.626	1.173	*	2		90	1.236	-1.704	1.701				2.430	5	3.788	0.0	2.442	9	1.526	2	3	35:	- 22	205	010	. 178	<b>5 2</b> .	.5%	.732	1.137	•
		315									1 0 01		3	297	212	*:134	058	ê			376	**	2.0	:	;	2.50	5								_	_	_		_			٠.		_
	*	7	-2.12	\$ 7	0	2:	7:17	2.5		8.0			¥	1	-3.25	-2.15	*	Ĉ.		3.32	4.43	4.43	•	•	1	7 • E	1	14.4	2.5	4	9	4.32	3.25	2.10	1.63	5.	.57	• 92	**	-1.03	*:-	-2.03	-3.10	7.55
910	<b>34</b> :	53	~	3	3				•	E	2	•	<b>.</b>	1			15		į	ş	~	14				3	4	:	1	•	-	•	63	=	~	82	2	-	<b>~</b>	\$	1	<b>.</b>	•	<b>~</b>
N - , 4010		74. 136.									0454° × N		AB CAF		_	_			_		_					2006	AB CAF		_				_	_	_		_		_		_	_	_	
0100" - NB 026"	3	<u></u>	<u>:</u> :						2		700 RN 4230	;	<b>9</b>		2	Ęį		9	ŧ.	ŧ.	Ž.		582			DOCC - NH OCO.	•	140		•		2.	22:	.173	5.	٥.	<u>\$</u>	=		.167	3	3.	3	.171
8 - 4.520 RM4010	Ca Ca8	1.899 .154	#: #:	191	412	41.130	1712	-2-205 .103	1111 200	-2.324 .102	G = 7.700 AM = .4230		<b>1</b> CM CM	116. 540.1	1.206			101	245 -299	·	1.100	1.000	502 502			DOCC - NA DOCC - D	CM CAB	-3.682 .190	-3.053 .188	-2.491 .184	-1.986 .181	1.533	-1.114 .176		0.1.	545 .170	106 - 169	991. 000.	1175 .167	. 366 . 167	263	#	P1: -11:1	1.524 .171
010+" - NE 025" - 5 04" .	GA CN CAB	1 310 1.899 .154	124 144	107. 407. 100.	191 - 191 - 190 ·	A11. 021.12 11.17	200 -1-712	362 -2-265	244	.366 -2.324 .192	1.10 G = 7.700 AM = .4230	:	TO CH CAS	1 297 1.642 .313	217 1.206 .304	047 Phot 991-	20%	104. 471 550.		141 -144	762. 001.1- 622.	200° - 20	013205		4 4 4 4	DOCC - NY OCEC - D DOC	CN CH CAB	.728 -3-682 -190	. 607 - 3.053	400 -2.401 .184	181. 400 -1- 504.	011 -1.533 .1.0	.229 -1.114 .176	.153 746 .173	011 948 110	071 770.	106 169	991. 000. 000.	191. 611. 960-	074 .368 .167	201. 696. 611	152 .761 .166	220 1-110 -160	309 1.524 .171
010+* - MR 056.4 + \$ 00 M	GA CN CAB	1.899 .154	124 144	107. 407. 100.	191 - 191 - 190 ·	A11. 021.12 11.17	200 -1-712	362 -2-265	244	.366 -2.324 .192	M = 1.10 G = 7.700 AM = .4230	:	<b>1</b> CM CM	1 297 1.642 .313	217 1.206 .304	047. Bear 991.	20%	104. 471 550.		141 -144	762. 001.1- 622.	200° - 20	013205		4 4 4 4		CM CAB	.728 -3-682 -190	. 607 - 3.053	400 -2.401 .184	181. 400 -1- 504.	011 -1.533 .1.0	.229 -1.114 .176	.153 746 .173	011 948 110	071 770.	106 169	991. 000. 000.	191. 611. 960-	074 .368 .167	201. 696. 611	152 .761 .166	220 1-110 -160	309 1.524 .171
0100, - MR 052.6 - 8 00, - M	ALP CN CN CAS	1 310 1.899 .154	-2.11124 .716 .144	1910 ADV - 160 - 101	191 - 215 - 880 - 91 · 1	2.17 .106 .1.130 .174	0.1. CIT-1- 00C. 45.6	4.32 - 2.26 - 4.39	0.00 ELL. 1- 445. BA.4	4.32 .300 -2.324 .102	.4220 H = 1,10 G = 7,700 AM = .4230		TO CH CAS	-4.32297 1.642 .311	905. 12.5 27. 1.206 . 304	067° 000' 001° C1°7°	202 101 102	106. 671. 250. 60.1	1.10 .064245 .299	6A. (94 141. B1.5	762. 001-1- 622. 72.6	900° 660°1- 962° 66°4	202 - 202 - 10°		4 4 4 4	·	CN CH CAB	8.76 .728 -3.682 .190	7.62 - 607 - 3.053 . 188	4.57 .446 -2.481 .184	181. 400.1. 504. 54.5	PC1 - 10.53 - 1.533 - 1.78	.229 -1.114 .176	2.19 .153746 .173	0.1. 40 +11. 54.1	071. 686 770. 00.1	•91· •01·- 0K0° •5°	991. 000. 000. 10.	191. 175036 .175	1.00076 . 368 .167	901. 696. 611 66.1.	-2.12152 .761 .100	-3.21220 1.119 .16d	-4.30304 1.524 .171
•	ALP CN CN CAS	-4.25310 1.899 -154 -3.10211 1.252 -140	201 -124 -126 -144	25. 440. 100. 100. 10. 10. 10. 10. 10. 10. 10.	EAL. 512. 000. 01.1	ACT 100 11 100 100 11 100 11 100 11 100 11 100 11 100 11 100 11 100 11 100 11 1	3.24 .24.1. 00.	.392 -2-205 -103	200 - 111.1. A.S. A.A.	.426 -2.32 .366 -2.324 .102	N = 1.10 G = 7.700 AN = .		ALP CN CN CAB	116. 5401 705 56.4	406. 907.1 1.2. 62.6	one boot sells of the sells of	500. 601. 110.0 60.	101 421 400 401	1.16 .064245 .299	64. 644. [41. 81.5 [84.	762. 001-1- 622. 75.6	906° 66°1- 962° 66°0 06°°	202 - 202 - 10°			·	CN CH CAB	.300 8.74 .728 -3.682 .190	30.5 - 30.6 - 3.053 - 3.053	. 307 6.52 .499 -2.491 .184	181. 400.1. 504. 54.5	0.300 0.34 0.35 -1.533 0.170	3.26 .229 -1.114 .176	611. 84 181 - 183 - 174	010 996 - 110 011 110 0110	071. 686 770. 90.1 AIE.	991° 90°° 98°° 216°°	991. 000. 000. 10. 11E.	791. \$11. 030 - 25. 114.	.314 .1.04074 .346 .167	910 . 100 . 1113 . 100	.317 -2.12152 .761 .100	.320 -3.21228 1.119 .168	.314 -324 -304 1.524 .171
5.700 Rh3616 M -	CAF CN CN CAB	-404 -405 -405 -310 1-899 -154 -403 -403 -3140 -3211 1-252 -140	-162 .467 -2.11 -2.15 .746	251 407	190 - 011 - 190 OIL	ACTION CONTRACT CONTR	STATE OF THE PROPERTY OF THE P	100 200 200 200 200 200 200	-100 -100	. 187 . 426 4.32 .346 -2.324 .102 .	N = 1.10 G = 7.700 AN = .		CAF CN CN CAB	116. 5401 705 56.4	906. 907.1 712 62.6. 954. 916.	one man and the state of the st	200 COL 110.0 CO. 12.0 CT. 12.0 CT.	CG. GC1 550. GO.!. GC4. G16.	.332 .481 1.10 .064245 .299	842. £44 [41. B1.5 £84. £84.	793. 001-1- 622. 75.6	900° Cho*!- 962° CC** OS** ACC*	202 203 101 101 202 203		000 to 00	• B M M OCO*C B O 0007 B M	D CAB CAF ALP CN CAB	.300 8.74 .728 -3.682 .190	1 .227 .309 7.62 .607 -3.053 .168	22.2 . 90.7	-210 -100 p. 100	1216 .309 6.34 .313 -1.533 .170	3.26 .229 -1.114 .176	641. 844 881. 91.5 016. 705	1204 .314 .344 .340 .370	off. 68t. 170, 90,1 Alt. 505.	991 302 315 38 38 38 38.	991. 000. 000. 10.	101. 203 -1032031036 -175 -167	.203 .314 .1.04074 .348 .167	901. 695. 611 96.1.	-202 -317 -2-12152 -761 -100	P91 - 9171 0-25- 12'5 926 1-110	.208 .319 .4.30309 1.524 .171
	CP CAB CAF ALP CN CN CAB	. 180 .404 .525 .510 1.899 .194 . 170 .403 .518 .521 1.252 .146	1.013 .182 .407 -2.11124 .716 .144	THE PART TARE PROTECTION TO SERVE SERVE SERVE	THE CASE SECTION OF THE CONTRACT OF THE CONTRA	-1-22: 182 183 183 174 175 175 175 175 175 175 175 175 175 175	-1-75 -175 -175 -175 -175 -175 -175 -175	ののは、 はののでは、 なのので、 なのので、 なんだっ あるだった!	-2-050 -180 -180 -120 -120 -120 -120 -120 -120 -120 -12	30+ -107 -420 4-32 -300 -2-324 -102	9 = 7,440 RN = .4220 N = 1,10 G = 7,700 RN = .		CAS CAS ALP CAS CAS	116. 5461 795 - 56.4- 000. 066. 1	906. 907.1 712 62.6- 954. 916. 916.1	DATE STORY GROWN STORY STORY STORY	THE TAXABLE PARTS OF THE TAXAB	MORE CARE TRANSPORT TO THE STATE OF THE STAT	1010 .000 100 0101 1010 .000 1000 1000	1-1.000 .335 .483 .248 .141643 .209	795. 001-1- 625. 75.6	SORT NECTT SERVICE SERVICE OF COMMUNICATION	562" 502" 810" 10" 00" 00" 00" 00"		0001 - 44 9004 - 4	· BEN OCHIC B D OO'S B M	CAS CAF ALP CN CM CAS	001. 284.1- 857. 45.8 .300 .3.582 .3.90	-9-16-2 -227 -305 7-5-2 -601 -9-059 -188	-2-570 -222 -30? 0.45 -2-49 -2-491 -184	-2.021 .210 .300 5.43 .402 .143	-1-9-50 -10-	-1.000 -1.01 -312 3.20 -220 -1.0114 -176		0.1. 46. 41. 54.1 1.4. 40.2. 114.	1,04 .202 .310 1.04 .077349 .170	184 187 184	-02. 150- 10. 111. 150- 150- 150- 150- 150- 150- 150- 150-	**** **** **** **** **** **** **** **** ****	194 996 970" 90"1" 195 666"	- 503 - 503 - 104 105 105 106	-202 -302 -317 -2.12152 -761 -100	1.078 .203 .320 .3.21228 1.119 .168	1.902 .208 .319 .4,30309 1.524 .171
0 - 1.700 Ah3616 H -	SE CH CH CAB CAF ALP CN CH CAB	1.057 .100 .404 -4.25310 1.099 .154 1.401 1.403 .194 1.34	-195 1.013 .182 .467 -2.11124 .716 .144	TOTAL POOL POOL POOL POOL POOL POOL POOL PO	CANAL CONT. CONT. CANAL CONT. CONT. CONT.	4201-23: 184 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -1-14 -14	200 - 100 -	MONTH WERE CONTROL CON	916 - 2160 - 180 -	.035204 .187 .424 4.32 .346 -2.324 .102	0 RM4220 M - 1.10 G = 7.700 AM		CAS CAS ALP CAS CAS	300 1.679 330 450 45.32 297 1.642 311	+000 - 902-1 L12 K2-K- +54- +1K- +1K-1 +22	DATE DOT SOLL CITY SALE OF THE	TANK DATE THE CO. THE TANK THE	100 001 101 100 100 100 100 100 100 100	193 - 193 - 193 - 193   1010 - 1040 - 2040 - 2040	. 218 -1.090 .339 .483 .2.18 .141443 .299	795. 001:1- 825. 75.8 15+. 588. +06:1- 495.	SORT NATION OF THE TOTAL T	567' 507' ETO" 10' 084' CTC CTC CTC' 504' C40''		1	* PR 060*C P 00*7 P K	CN CN CAS CAF ALP CN CN CAS	-3,794 .231 .300 8.74 .728 -3.682 .190	.ele -9.162 .227 .305 7.62 .607 -9.099 .166	+303 -2+570 +222 +30? a.52 a440 +2+401 +184	-401 -2-021 -219 -309 5-84 1-402 -11-086 -181	. 310 -12-546 -12-6 -304 -12-64-6 -12-64-6 -12-64-6	475 -1.000 -1.000 5.20 5.20 15. 115. 15. 17. 17. 17. 17. 17. 17. 17. 17. 17. 17	.147 404 - 207 .310 .2.19 .153 746 .173		01: -84770. 1.04 1.04 -011.	-030 140 - 202 100 100 100 100	.000 .000 to. 1311 .000 .000 .000 .000	1036 104 103 1036 1036 104	-071 394 -203 314 -1.04074 394 3167	- 101 - 202 - 201 - 401 - 401 - 402 - 401		222 1-076 -203 -320 -3-21226 1-119 -166	-,309 1,902 ,208 ,319 -,4,30 -,369 1,524 ,171

Table 51. Aerodynamic Coefficients for Configuration BlT13SX5

.4190	CAF	376	40	7	-		?		200	.336	-		.3860	34.5		.380	.384	.347	96.	990	. 369	387		36.0	386			.3100	CAF	28.5	234	.231	. 233	-236	• 52*	.249	. 233	•534	• 236	.241	.234	.237	.234	.236	.230	.236	
z	CAB	.323	562.	-288	687		205	356	. 341	8	92.5	·82•	1 2		9	\$220	.219		90.	2	500	•219	277		502	ì		a N	5	:	.122	121	.120	120	.120	.119	.119	-118	.119	.119	.119	.119	27.	.119	2:	27.	
7.090	5	1-670	2695	•235	- 264	647	924	1.370	1.807	2.931	8	.192	8.560	;	5	1.711	1.218	.752	.353	060	510	996	1.458	24.	3.100			4.130	5		3.241	-2.535	1.951	-1.456	-1.047	664	470	289	127	.063	-239	.379.	\$65.	.790	1.185	1.575	
. 0		286						•	•	•			1.50 0 .1	į	5	312	222	137	059	.023	.103	198	- 277					. 00	Š	•	652	. 521	904	308	. 223	. 142	200	290*	•050	012	050	002	126	166	247	329	
	ALP.	-4.27	-2.12	-1.05	.03	01.1	2.17	3.26	<b>4.33</b>	6.50	1.10	-1.04		:	1	-4.38	-3.27	-2.17	-1.07	ģ	*: :-	2.24	3.35	4.4	2.5	5			ALP	:	7.37	6.23	97.5		3.04	1.97		•	36	91	49.	-1.5	7	-2.27	-3.35	4.37	•
060+	CAF	.356	.375	.385	.384	.377	.364	.353	.337	.320	.376	.390	. 41 60		<b>4</b>	.438	*	2++5	244.	.438	643	164.	924.	\$75	11,	. 45		.3100	CAF		254			386	9	246			190	,,,	24.5	26.36	250		55.	25.	
ž	5	.234	.213	82.	\$2	602	•12.	.233	<b>2</b>	.285	.213	<b>8</b>	ž		CAB	2,50	243	.237	.239	.242	.243	.255	.264	.27	2	-535		2	CAB		5				1		1						7		1		!
6.750	5	2.039	90	.423	055	576	-1.177	-1.657	-2.137	-3.346	593	•*	.400		5	1.729	1.23	702	351	076		942	-1.463	-2.000	-3.30	• 0 • 0		4.930	5		-3.910	R :	016.7-				2 4 4	17.			3				7		1.73
.0	3	325	*	063	•10.	901.		.275	.351	.556	.102	1.04	.30		₹	- 300	212		150	0.00	103	.185	.275	.367	.583	•055		.50	₹		£:			2							200						314
	4	-4-27	21.2-	-1.05	• 05	1.04	2.17	3.24	4.32	.4.4	1.0	-1.05	=		ş	*				50	1	7.7	46.4	\$.	4.4	\$0.			ALC		1	£ :	•	•		7.6				7	8:		•	3	21.5	2.5	
010	CAF	.344		2	96.	141	940						92.79	<b>?</b>	CAF									9				3500	*		.277	2.5				4		281		280			900		.288	4	1
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6.520		2.232											7.700	3	•													5.650	8																		
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	_	***	-3.19	-2.12	60.1.	N 60	3		5			•			• 10		7.7	-3.22	-2.14	90.7	20.	2:5	3.26		•••				*								::		7		60.		-		200		
3413	<b>**</b> 5	.354									400	• 25.	,	077.	CAF		264.	05.	•			\$4.		17.				9800		į	. 04.	.00.	. 305	.310	.313	.3.	.315	.319	.315	.31	.304	.310	.311	.313	.319	.317	
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Table 52. Aerodynamic Coefficients for Configuration B1T14X1

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ž	CAB		įį	1			9	ž	2			:		*		CAB	•	261.			481	187		761	. 199	+12.	. 193			*	į	CAB	3		:	100	108	901.	104	- 102	. 102	. 102	101	101		===	9	103		
1.090	5		1.643	.132		1350	206			997.7	2.4			045.4	;	5		101-1	26.	R .		288	703		995-1	2.681	1.053			9		5	;	-3-132	7	76.2	-1.331	920	546	372	222	103	.019	.137	.251		410	176		
9 00			296											9	2	Š		-, 252	155	560	7000	3	5	761	925		1	:		8	3	3					,0	217	5	660	.059	.027	•00•	036	067	- 105		902		
H = 1.00			-3.21			_				_		_		1		ALP		-4.37	-3.25	-2.15	-1.05	5	61.9	2.2		,				•		A.P.	i	. 20		26.0	7.63			15.1	÷	4.	•00	9	-1.13	-1.65	11.2-	C2.5.	7	
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06.04	CAF	,	.264 .267	.267	992.	562			167		152	607		,	8	CAF		•52•	.258	200	607	707	10.	167.			2	.271			3100	CAF		.231	.222	. 229	•223	2	667		213	233		.233	.232	.226	.352	-229	. 22.	
		;	.112	.110	•10	•10•	9:	2:	*:	171	61.	:::		į	E E	CAB		. 195	.193	061.	120	.165	981	.189		35	027	183			z	840		.142	.143	.142	5	.133	2	921			2		0.1	0.1	٤.	.127	.127	
6.750		;	1.922	.786	.274	.279	1		-042	.571	.621			•	•	ă	•	161	.637	505	.203		. 360	969	60.	:	2.53	500	1		4.930	5	•	3.423	2.721	2.161	1.060	1.569	1.226	.765						3		1.250	1.503	
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Table 53. Aerodynamic Coefficients for Configuration B1T14X2

8	4	****	2.2	*	252	.269	. 3860	S.	-285	£.	286	286	.27	273	.276		.3100	CAF	27.	. 233	.227	.23	***	.222	210	.222	122.	17.	217	. 513. 12.	-512
=	CAB	14.5	25	291.	•	171		8	191	261	261	ij	8	22	<b>.</b>		=	<b>88</b>	-101	90	100	501.	6	101	ě	.103	201•	200	607	225	.103
7.040	5	1.53	-029	1.044	*11.5	.00	. 360	5	1.10	395	910-	0+2-	1.027	-1.527	•11.		4.130	5	3.862	2.133	1.923	909-1		2 4	263	• 100	610	£ 7.	516	4. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1.	1.590
8.1		126.					.50	5	*	90.		290	į		- je		3.00													951:-	
- - -	AL	27.5	20.	2.18	4.35		•	ALP	4.4	-2.12	90.	1.15	3.37	**	-3.24			4.4	5.52		2.50	4.21	3.5	10.5	8	7	6	16.1	-1.62	-2.14	12.4-
04.0	<b>₹</b> ₹	22.25	2.7	7.00	7.		314.	<b>1</b>	.274	.287	.202	276	.276	.273	.282		.3100	CAF	.224	.222	•22•	122	.221	.224	.223	922	224	.225	27.	22.	
2	<b>649</b>	***		===	137	193	=	3	.203	6	=	*	Ŕ		.197		ž	5	# 3	.145	?:	2		.132	=	2 2	12	2			
. 6.750	. 5	1.044	200	-1-200	***	-3-736		5	1.657	Ē	.020	***	-1.070	-:->. -:->.	1.204		. 4.930	5	-3.636	-2.102	000	-1.56	2.0		 8	2530	241	3.	. 573	1.240	
•		55.5					1.30	3	289	2	6.0	3	22.	.323	702*-		2.50 0	3	* 4	*	-361	9		2.	•0.	į	200		*		***
	3	2.2.2.	20.	2.2		-2.13		***	7	-2-13	\$0. 1	*:	**		-3.22		•	\$	57.5	6.53	5.45		2.2	1.68	1.14	7.	44		-1.50	67.	-
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The static stability char	racteristics of	ring tail	ls on
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of wind tunnel tests. Mach	numbers varied	l from (	). 8 to
4.5, and ring diameters var	ied from 1.25	to 2.50	calibers.
This report presents a tabul			
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